FAA APPROVED MODEL LIST (AML) NO. SA00096SE

AEROCET, INC.
for
AEROCET 206 CARGO PACK INSTALLATION

ISSUE DATE: June 28, 2006

ITEM	AIRCRAFT MAKE	AIRCRAFT MODEL	ORIGINAL TYPE CERTIFICATE NUMBER	CERTIFICATION BASIS FOR ALTERATION	FAA APPROVED AIRPLANE FLIGHT MANUAL SUPPLEMENT		FAA APPROVED MASTER DRAWING LIST		AML REVISION DATE
					Number	Revision*	Drawing	Revision*	
1.	Cessna	206, P206, P206A-E, TP206A-E, U206, U206A-G, TU206A-G	A4CE	CAR 3	A-10012	Revision 1, dated 6/19/06	10-10005N-1	Revision B, dated 8/5/05	
2.	Cessna	206H	A4CE	Part 23	A-10016	Revision I/R, dated 6/19/06	10-10005N-1	Revision B, dated 8/5/05	
3.	Cessna	T206H	A4CE	Part 23	A-10017	Revision I/R, dated 6/19/06	10-10005N-1	Revision B, dated 8/5/05	

* Or later FAA Approved Revision

FAA Approved: _____

Acting Manager, Seattle Aircraft Certification Office

REV	DATE	DESCRIPTION	BY	CHECKED
A	10/22/99	DRAWING 10-10010 UPDATED UPDATED PROCESS SPECS	TH	TH
B	8/5/05	DRAWINGS 10-20010 & 10-20012 UPDATED REMOVED WORK ORDERS FROM MDL	RB	TH
C	4/21/06	ADDED SHEET 2. REVISED 10-10011, 10-10012, 10-20010, 10-20012 10-20015, 10-20017 & 10-20020; ALSO A-31005, & A-10014. ADDED 10-20021, 10-20022, 10-20027, 10-20028, 10-20029, 10-20030; ADDED AFMS' A-10012, A-10016 & A-10017; AND ADDED A-10033.	RB	TH
D	6/2/09	REVISED DOCUMENT A-10014 & 10-20021.	RB	TH
E	11/22/10	REVISED DOCUMENT 10-10010, ADDED THE FOLLOWING DRAWING TO MDL, 10-10015, 10-10016, 10-10030, 10-10031, 10-10032, 10-10034, 10-10035, 10-10037, 10-10038, 10-T10038, 10-10039, 10-10040, 10-10051, 10-10053, 10-10054, REVISED DOCUMENT A-31005, CHANGE (THIS DOC.) SHEET 2 SIZE WAS A SIZE IS B SIZE 10-10005.	RB	TH
F	3/22/12	REVISED DRAWINGS 10-10010, 10-10030, 10-10034, 10-10035, 10-10037, 10-10040, 10-20015 & UPDATED A-10033 TO SHOW REV 1 (PREV. APPROVED). CORRECTED TYPO FOR 35-80004 TO REV A. DATE WAS CORRECT.	RB	

(SEE SHEET 2 FOR LISTINGS)

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TOLERANCES: EXCEPT WHERE NOTED

TOLERANCING AND DIMENSIONING PER ANSI Y14.5M

LINEAR
.X = ± NA
.XX = ± NA
.XXX = ± NA
ANGULAR: ± NA DEGREE
FRACTIONAL: ± NA
FINISH REQUIREMENTS: NA RMS

REMOVE ALL BURRS AND EDGES .010-.015
CONCENTRICITY: NA
RUNOUT: NA
PERPENDICULARITY: NA
PARALLELISM: NA
INTERNAL CORNER RADII: NA
DIMENSIONS IN INCHES
DIMENSIONS APPLY BEFORE PLATING

DESCRIPTION:

206 CARGO POD MASTER DRAWING LIST

AEROCET

Priest River, Idaho

FILE NO.	DRAWN BY:	DATE:	SCALE:	DWG NO.	REVISIONS
POD\10-10005	TH	10/22/99	NONE	10-10005	F
LOCATION	APPROVAL	DATE:	SIZE	USED ON:	SHT.
harddrive	TH	10/22/99	A	STC SA0096SE	1 of 2

DRAWING #	DATE	REVISION	AIRCRAFT APPLICATION
10-10010	**3/1/12**	**C**	**ALL**
10-10011	8/24/05	A	N/A
10-10012	8/26/05	A	ALL
10-10015	5/5/10	I/R	ALL
10-10016	5/10/10	I/R	ALL
10-10030	**3/1/12**	**A**	**ALL**
10-10031	11/22/10	I/R	ALL
10-10032	**6/16/11**	**A**	**ALL**
10-10034	**6/16/11**	**A**	**ALL**
10-10035	**6/17/11**	**A**	**ALL**
10-10037	**6/16/11**	**B**	**ALL**
10-10038	11/22/10	I/R	ALL
10-T10038	6/9/10	I/R	ALL
10-10039	3/31/09	I/R	ALL
10-10040	**3/1/12**	**A**	**ALL**
10-10051	5/10/10	I/R	ALL
10-10053	5/10/10	I/R	ALL
10-10054	6/14/10	I/R	ALL
10-20010	2/11/05	A	EXCLUDING H MODELS
10-20012	2/11/05	A	EXCLUDING H MODELS
10-20015	**3/14/12**	**B**	**EXCLUDING H MODELS**
10-20017	8/25/05	A	*ALL
10-20020	8/26/05	A	ALL
10-20021	6/2/09	A	T206H
10-20022	3/18/06	I/R	206H & T206H
10-20027	3/18/06	I/R	ALL
10-20028	3/18/06	I/R	ALL
10-20029	4/20/06	I/R	T206H
10-20030	4/20/06	I/R	206H
35-80004	11/22/10	A	ALL

DRAWING #	DATE	REVISION	AIRCRAFT APPLICATION
A-31005	5/27/10	G	POD CONSTRUCTION ONLY
INSTALL DOC.			
A-10014	3/10/09	2	ALL
AIRPLANE FLIGHT MANUAL SUPPLEMENTS			
A-10012	4/20/06	1	206, P206, P206A-P206E, TP206A-TP206E, U206A-U206G, TU206A-TU206G
A-10016	4/20/06	NONE	206H
A-10017	4/20/06	NONE	T206H
SERVICE MANUAL AND INSTRUCTIONS FOR CONTINUED AIRWORTHINESS			
A-10033	**3/22/12**	**2**	**ALL**

*PER INSTALLATION DOCUMENT A-10014. 10-20017-1 IS NOT APPLICABLE TO H MODEL AIRCRAFT.

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TOLERANCES: EXCEPT WHERE NOTED

TOLERANCING AND DIMENSIONING PER ANSI Y14.5M
LINEAR
.X = ± N/A
.XX = ± N/A
.XXX = ± N/A
ANGULAR: ± N/A DEGREE
FRACTIONAL: ± N/A
FINISH REQUIREMENTS: N/A RMS

REMOVE ALL BURRS AND EDGES .010 .015
CONCENTRICITY: N/A
RUNOUT: N/A
PERPENDICULARITY: N/A
PARALLELISM: N/A
INTERNAL CORNER RADII: N/A
DIMENSIONS IN INCHES
DIMENSIONS APPLY BEFORE PLATING

DESCRIPTION:

206 CARGO POD MASTER DRAWING LIST

AEROCET

PRIEST RIVER, IDAHO (208) 448-0400

FILE NO. POD/10-10005	DRAWN BY: RB	DATE: 4/21/06	SCALE: 1/1	DWG NO. 10-10005	REVISIONS F
LOCATION harddrive	APPROVAL TH	DATE: 4/21/06	SIZE B	USED ON: STC SA0096SE	SHT. 2 of 2
	REV. APPVL	DATE: 3/22/12			

INSTALLATION MANUAL

FOR

AEROCET CARGO PACK

ON CESSNA 206 SERIES AIRPLANES

This Installation Manual is a necessary part of an FAA Approved data package for STC #SA00096SE, providing requirements for the acceptable installation of the Aerocet 206 Cargo Pack.

LOG OF REVISIONS PAGE

REV.	PAGES AFFECTED	DESCRIPTION	FAA APPROVED	DATE
I/R	ALL	Initial Release		
1	ALL	Added Cover Page Style change to Title Block, updated Company address Added Section "G" Added Figure "6" Updated Parts Listing Updated Figures 1-5 with current hardware and specific model call-outs with regard to Models 206H and T206H		4/20/2006
2	ALL	Changed Rivnut part number, NAS1329S3K130, was NAS1329S3K140, page 3, PART LISTING, and Page 11, FIGURE 3, illustration		3/10/2009

CARGO POD C-206

KIT PARTS LIST

QUANTITY	PART NUMBER	PART NAME
1	10-10010	Pod Shell
32	MS27039-1-13	Screw
32	MS35335-32	Washer
32	NAS1149F0332P	Washer
20	NAS1329S3K80	Rivnut
7	NAS1329S3K130	Rivnut
4	NAS1329S3K180	Rivnut
10	NAS1329S06K120	Rivnut
20’	R860064	Seal
1	10-20017-1	Placard
1	10-20017-2	Placard
1	10-20022	Placard
1	10-20029	Cargo Pack Placard Cessna T206H
1	10-20030	Cargo Pack Placard Cessna 206H
1	1213509-1 (Cessna Part)	Baffle-Inbd-LH (thru U20601604)
1	1213509-2 (Cessna Part)	Baffle-Inbd-RH (thru U20601604)
1	1213509-3 (Cessna Part)	Baffle-Outbd-LH (thru U20601604)
1	1213509-4 (Cessna Part)	Baffle-Outbd-RH (thru U20601604)
1	10-20010-L *	Baffle-Outbd-LH (thru U20601605 & on)
1	10-20010-R *	Baffle-Outbd-RH (thru U20601605 & on)
1	10-20012-L *	Baffle-Inbd-LH (thru U20601605 & on)
1	10-20012-R *	Baffle-Inbd-RH (thru U20601605 & on)
2	10-20015 *	Link
1	10-20027**	Fairing
1	10-20028**	Fairing
1	1200641-13 (Cessna Part)	Fairing
1	1200641-14 (Cessna Part)	Fairing
2	10-20020	Seal

* Not necessary for 206H and T206H Models

** MK206-53-01-2 is an alternate part number for 10-20025. 1200641-13 and 1200641-14 are alternate part numbers for 10-20027 and 10-20028.

CARGO POD C-206

KIT PARTS LIST (Cont'd)

QUANTITY	PART NUMBER	PART NAME
4	MS21069-L3	Nutplate
4	MS35207-261	Screw
12	AN525-1032R6	Screw
10	MS35206-230	Screw
12	MS21042-3	Nut
2	AN316-4R	Nut
4	NAS1329S3K230	Rivnut
2	¼" Stainless AN742-10 *	Clamp
2"	306-4 Aeroquip *	Low Pressure Hose
7"	¼" OD 5052-0 *	Aluminum Tube
1	MS35489-7 *	Grommet
1	10-20025 **	Exhaust Extension T206H "Kit"
1	MK206-53-01-2 (Cessna Part) **	Exhaust Extension T206H "Kit"

* Not necessary for 206H and T206H Models

** MK206-53-01-2 is an alternate part number for 10-20025. 1200641-13 and 1200641-14 are alternate part numbers for 10-20027 and 10-20028.

CHANGE IN WEIGHT AND BALANCE

WEIGHT INCREASE	35.0 POUNDS	(NET CHANGE)
ARM	51.0 INCHES	
RESULTANT MOMENT	1785.0 POUND-INCHES	
INDEX	1.632	

1. **DESCRIPTION OF INSTALLATION**

A. (Refer to Figure 1) Installation of this kit consists of:

1) Removing two existing nose gear access panel assemblies and modifying them according to Figure 2.

2) Modifying auxiliary fuel pump vent if pump installed in fuselage area.

3) Positioning cargo pack, drilling attaching holes, and installing rivnuts in fuselage skin.

4) Attaching cargo pack to fuselage with screws.

5) Modifying cowl flaps by adding baffles and/or extensions (some come modified).

2. **INSTALLATION INSTRUCTIONS**

A. (Refer to Figures 1 & 3) Installation of Nose Gear Access Panel Assemblies.

1) Remove existing panel assemblies and modify according to Figure 2. Prime with zinc chromate. Using existing screws, install modified assemblies according to Figures 1 and 3.

NOTE: Modified panel assemblies will permit servicing nose gear strut without removing cargo pack.

B. (Refer to Figure 5) Modification of Auxiliary Fuel Pump Vent. (Applicable only to serial numbers U20601605 through U20601618 and U20601633 and on when auxiliary fuel pump is located in fuselage.) (NOT APPLICABLE TO 206H AND T206H MODELS)

1) Remove existing vent tube drain from aircraft.

2) Install tube (1), line (4) and grommet (5) with connecting hose (2) and clamps (3).

C. (Refer to Figure 3) Installation of Cargo Pack Assembly.

1) Determine centerline of pack assembly (7) and mark on outside of pack on the aft end. Centerline at forward end is determined by location of nose gear drag brace trough in pack assembly.

2) Mark centerline of aircraft on bottom fuselage skin across the station 100 area.

3) Before fitting cargo pack, loosen lower wing lift strut fairings and slide them about 12" up; lift strut out of the way.

4) Place pack assembly (7) under the aircraft and align cutouts (2, Figure 1) with landing gear springs. Position drag brace trough under nose gear drag brace. While pack assembly is being held in position, place a board crosswise under pack (7) and, using a jack, support the entire pack assembly firmly against fuselage. Be careful not to exert enough pressure to deform or spread edges of cargo pack.

5) Check centerline of pack assembly (7) with centerline marked on lower fuselage skin. Check position of existing holes locations on pod (5, Figure 1) in cargo pack with fuselage skin below the door sills. Holes should align with lower rivet line on both sides of aircraft. Adjust pack fore and aft so that holes fall between rivets as shown in Figure 1.

6) When pack assembly (7) is positioned correctly, drill thirty-one #10 (.193) holes through pack and fuselage, using hole locations scribed into the pack as a guide. **CAUTION:** Use a drill stop or some other device to assure that the drill does not go beyond the rivnut depth.

7) Remove pack assembly and enlarge #10 holes in fuselage to ¼" and cut slots as shown (use a Rivet key punching tool if available). Install Rivets (8), (10), and (11) in fuselage skin.

8) Apply seal (2) to cargo pack (7) holding in place with masking tape. If a sealer caulk is necessary, assure that it is non-corrosive to aluminum.

9) Attach pack assembly to fuselage using screws and washers (9). Use an ice pick to put holes in the seal for the fasteners.

D. (Refer to Figure 3) Installation of Seals and Fairings

1) Cement seals (3) to inboard side of fairings (5) and (12) using contact cement. Cut-out area in seals should be centered in relation to rectangular openings in fairings, with split end of seal aft.

2) Split aft ends of fairings to match split ends of seals, place fairings over landing gear springs, and position so bead at top of fairing matches bead edge of cargo pack. Using holes in fairings as a guide, drill five #26 (.147) holes in each side of cargo pack.

3) Remove fairings and enlarge #26 holes in cargo pack to #12 (.189). Install Rivnuts (6) in cargo pack and reattach fairings, using screws (4).

4) Clean fairings (5 and 12) and prime with zinc chromate primer.

E. (Refer to Figure 4) Cowl Flap Modification (Models 206, P206, U206 and Stationair) (NOT APPLICABLE TO 206H AND T206H MODELS)

1) Disconnect cowl flap controls (4) and install links (7) and checknuts (9). Retain existing checknuts (6) on cowl flap control as shown. Cessna supplied links (#1213626-1) can be substituted for (7).

2) With cowl flap control in closed position, adjust cowl flaps (by turning links) until cowl flap is 1.05” open at aft end. Openings to be measured between centerline of cowl flap at projected aft edge to lower firewall. Tighten checknuts (6) and (9).

3) Position baffles (1, 2, 3 & 5) with forward end of flange resting on flange of existing baffles, aft end of baffle aligning with aft end of existing baffle, and lower corner of angular cutoff aligning with upper aft corner of existing baffle. Use Cessna part # 1213509-1, 2, 3, 4 for aircraft serial numbers through U20601604. Use Aerocet part #10-20010-R, L and Aerocet part #10-20012-R, L for aircraft serial numbers U20601605 and on. Cessna baffles #1213687-1, 2, 3, 4 may be substituted for the Aerocet baffle parts #10-20010 and #10-20012.

4) Using holes in baffles as a guide, drill twelve #10 (.193) holes in existing baffles. Deburr holes and secure new baffles to existing baffles with screws (11) and (10). Options to substitute with rivets as shown in (11).

5) Adjust clearance of cowl flap baffles. Clearance between baffles and edge of cutout in engine cowl should be 1/8” (.125). Adjust by hand, bending baffles. Check for clearance with engine drain tubes.

6) Check cowl flap travel. Cowl flaps should be 1.05” open when control is in “closed” position and approximately 6.00” open when control is in “open” position.

F. (Refer to Figure 3) Cowl Flap Modification (Model TP206 & TU206) (NOT APPLICABLE TO 206H AND T206H MODELS)

NOTE: Baffles (1, 2, 3 & 5) are not required on turbocharged aircraft

1) Disconnect cowl flap controls (4) and install links (7) and checknuts. Retain existing checknuts (6) on cowl flap control as shown.

2) With cowl flap control in closed position, adjust cowl flaps (by turning links) until cowl flap is 2.50” open at aft end. Tighten checknuts (6) and (9).

3) Check cowl flap travel. Cowl flaps should be 2.50” open when control is in “closed” position, and approximately 8.00” open when control is in “open” position.

G. Model T206H Airplane Exhaust Pipe Extension Installation

1) For Model T206H, also install exhaust pipe extension as follows:

a) Using Aerocet Part #10-20021 or Cessna Part #1250999-1 Exhaust Extension as a template, drill four (4) Number 29 (0.136 inch diameter) holes in exhaust pipe.

NOTE: Extension must be flush with the exhaust pipe. It may be necessary to grind off approximately 1.5 inches of bead on exhaust pipe or reshape exhaust pipe in order to provide tight fit.

b) Attach extension using MS51861-34C Screws.

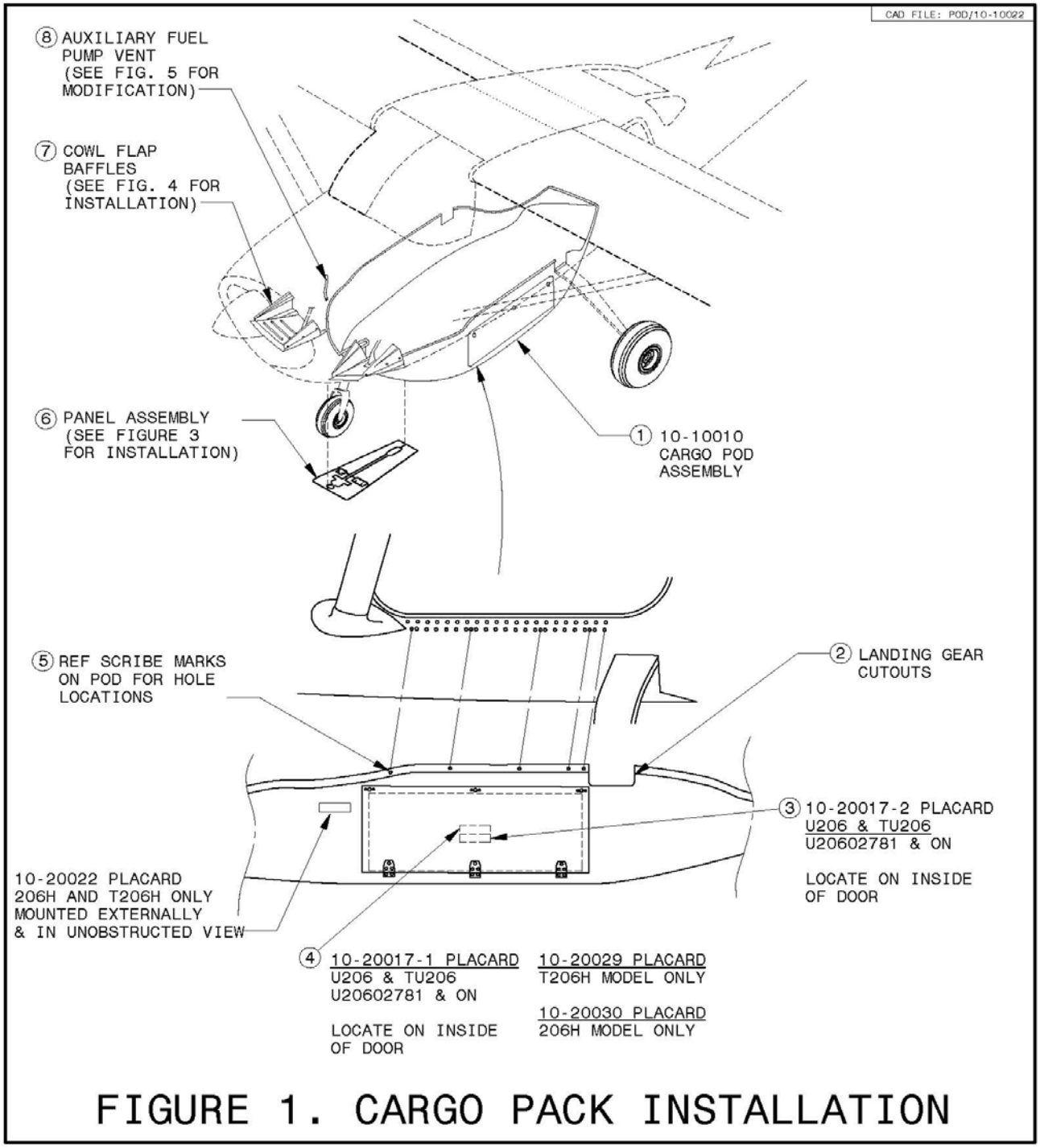
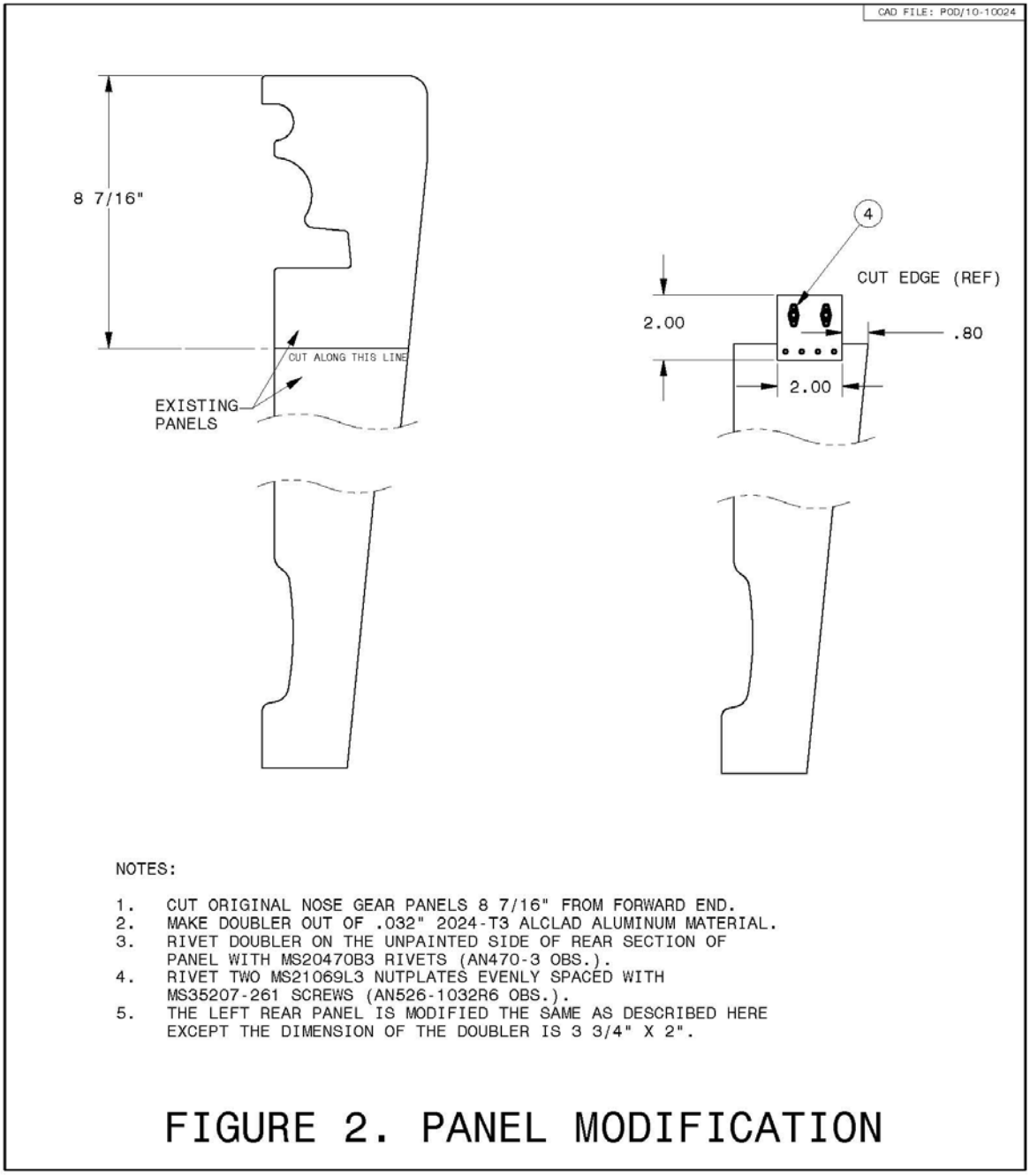

CAD FILE: POD/10-10022
⑧ AUXILIARY FUEL PUMP VENT (SEE FIG. 5 FOR MODIFICATION)
⑦ COWL FLAP BAFFLES (SEE FIG. 4 FOR INSTALLATION)
⑥ PANEL ASSEMBLY (SEE FIGURE 3 FOR INSTALLATION)
① 10-10010 CARGO POD ASSEMBLY
⑤ REF SCRIBE MARKS ON POD FOR HOLE LOCATIONS
② LANDING GEAR CUTOUTS
③ 10-20017-2 PLACARD
U206 & TU206
U20602781 & ON
LOCATE ON INSIDE OF DOOR
10-20022 PLACARD
206H AND T206H ONLY
MOUNTED EXTERNALLY
& IN UNOBSTRUCTED VIEW
④ 10-20017-1 PLACARD
U206 & TU206
U20602781 & ON
LOCATE ON INSIDE OF DOOR
10-20029 PLACARD
T206H MODEL ONLY
10-20030 PLACARD
206H MODEL ONLY

FIGURE 1. CARGO PACK INSTALLATION



NOTES:

1. CUT ORIGINAL NOSE GEAR PANELS 8 7/16" FROM FORWARD END.
2. MAKE DOUBLER OUT OF .032" 2024-T3 ALCLAD ALUMINUM MATERIAL.
3. RIVET DOUBLER ON THE UNPAINTED SIDE OF REAR SECTION OF PANEL WITH MS20470B3 RIVETS (AN470-3 OBS.).
4. RIVET TWO MS21069L3 NUTPLATES EVENLY SPACED WITH MS35207-261 SCREWS (AN526-1032R6 OBS.).
5. THE LEFT REAR PANEL IS MODIFIED THE SAME AS DESCRIBED HERE EXCEPT THE DIMENSION OF THE DOUBLER IS 3 3/4" X 2".

FIGURE 2. PANEL MODIFICATION

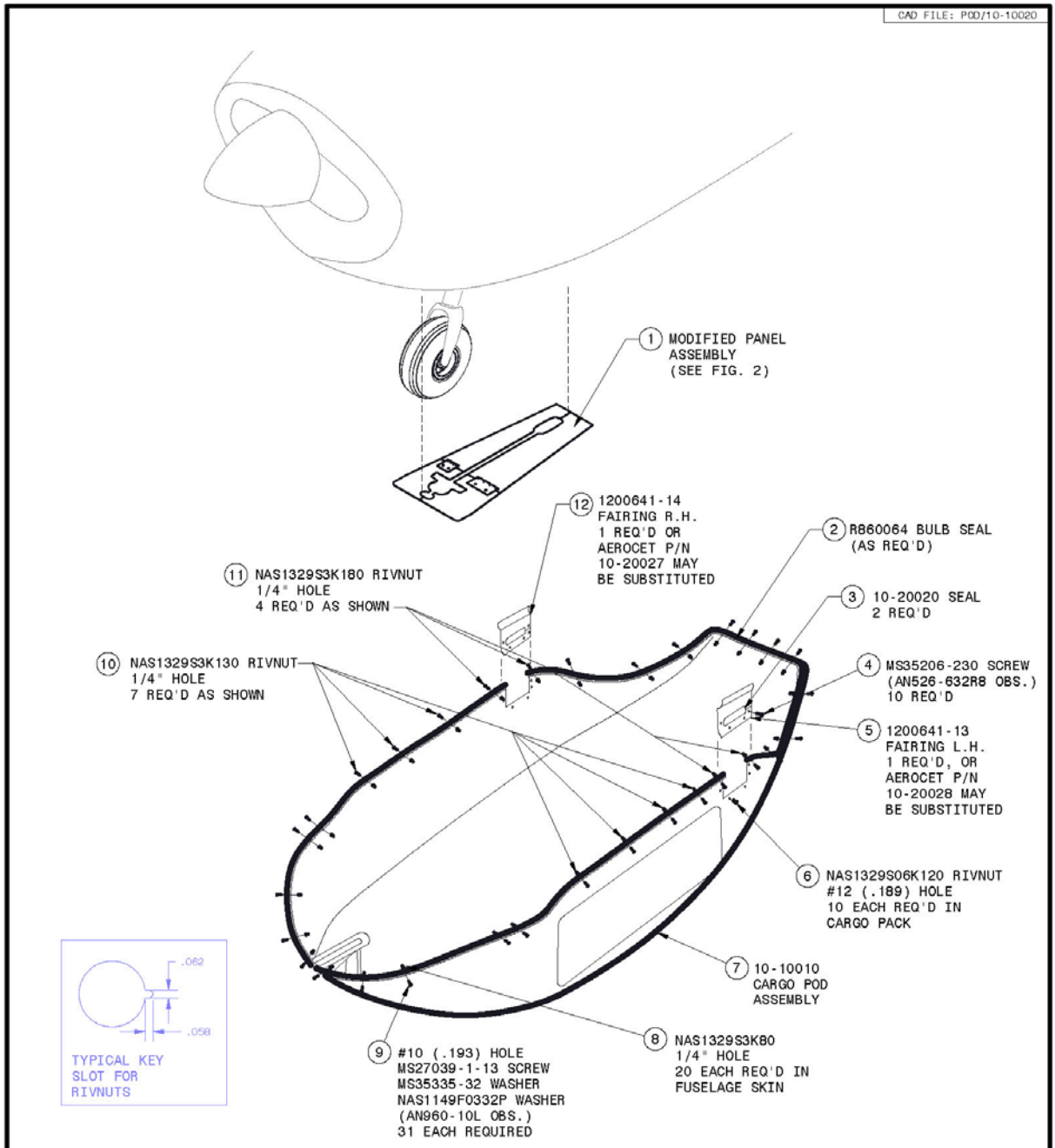

CAD FILE: POD/10-10020
1 MODIFIED PANEL ASSEMBLY (SEE FIG. 2)
12 1200641-14 FAIRING R.H. 1 REQ'D OR AEROCET P/N 10-20027 MAY BE SUBSTITUTED
2 R860064 BULB SEAL (AS REQ'D)
11 NAS1329S3K180 RIVNUT 1/4" HOLE 4 REQ'D AS SHOWN
3 10-20020 SEAL 2 REQ'D
10 NAS1329S3K130 RIVNUT 1/4" HOLE 7 REQ'D AS SHOWN
4 MS35206-230 SCREW (AN526-632R8 OBS.) 10 REQ'D
5 1200641-13 FAIRING L.H. 1 REQ'D, OR AEROCET P/N 10-20028 MAY BE SUBSTITUTED
6 NAS1329S06K120 RIVNUT #12 (.189) HOLE 10 EACH REQ'D IN CARGO PACK
7 10-10010 CARGO POD ASSEMBLY
8 NAS1329S3K80 1/4" HOLE 20 EACH REQ'D IN FUSELAGE SKIN
9 #10 (.193) HOLE MS27039-1-13 SCREW MS35335-32 WASHER NAS1149F0332P WASHER (AN960-10L OBS.) 31 EACH REQUIRED
.062
.058
TYPICAL KEY SLOT FOR RIVNUTS

FIGURE 3. CARGO PACK AND PANEL INSTALLATION

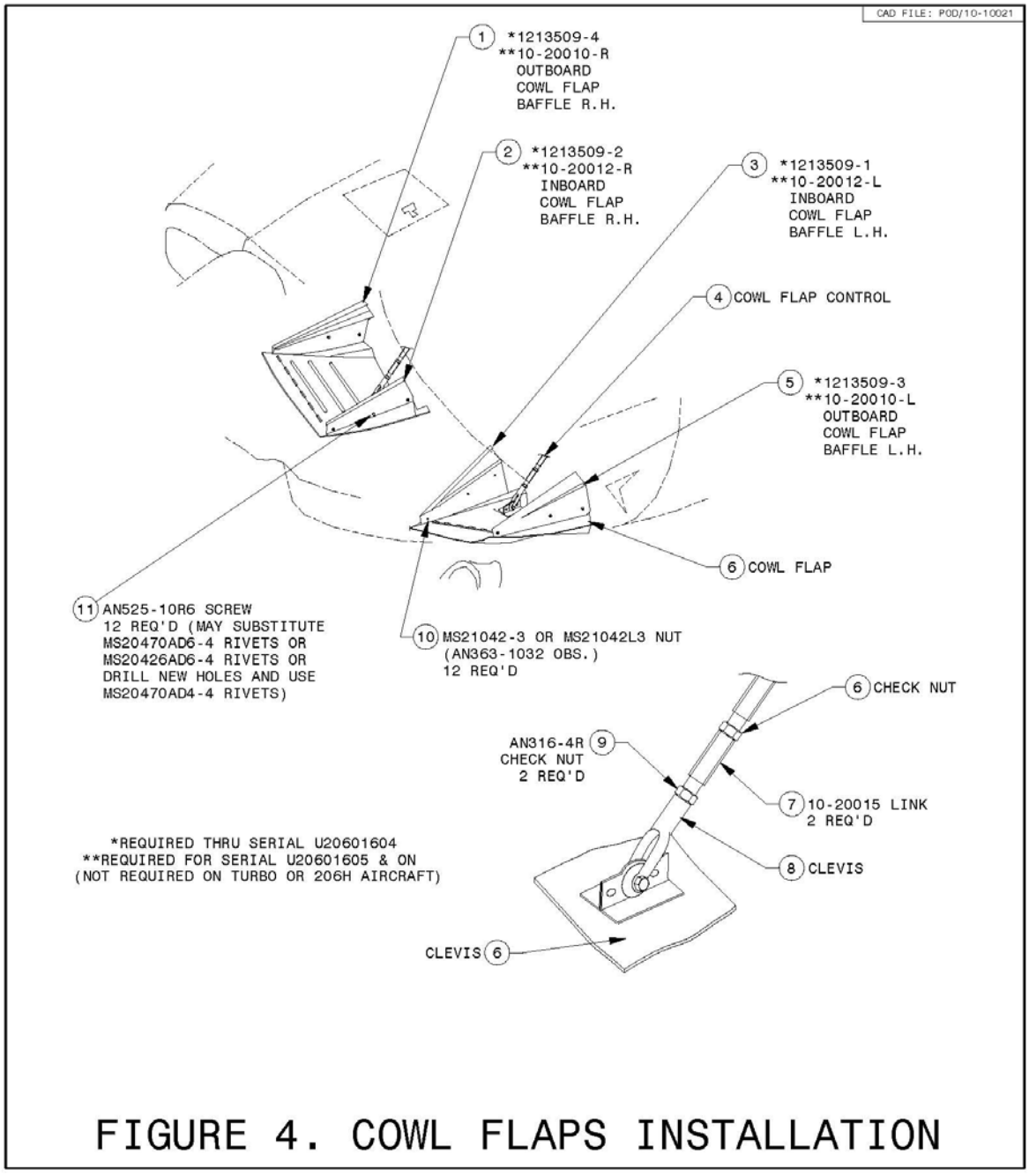

CAD FILE: POD/10-10021
1 *1213509-4 **10-20010-R OUTBOARD COWL FLAP BAFFLE R.H.
2 *1213509-2 **10-20012-R INBOARD COWL FLAP BAFFLE R.H.
3 *1213509-1 **10-20012-L INBOARD COWL FLAP BAFFLE L.H.
4 COWL FLAP CONTROL
5 *1213509-3 **10-20010-L OUTBOARD COWL FLAP BAFFLE L.H.
6 COWL FLAP
11 AN525-10R6 SCREW 12 REQ'D (MAY SUBSTITUTE MS20470AD6-4 RIVETS OR MS20426AD6-4 RIVETS OR DRILL NEW HOLES AND USE MS20470AD4-4 RIVETS)
10 MS21042-3 OR MS21042L3 NUT (AN363-1032 OBS.) 12 REQ'D
6 CHECK NUT
AN316-4R 9 CHECK NUT 2 REQ'D
7 10-20015 LINK 2 REQ'D
8 CLEVIS
CLEVIS 6
*REQUIRED THRU SERIAL U20601604
**REQUIRED FOR SERIAL U20601605 & ON
(NOT REQUIRED ON TURBO OR 206H AIRCRAFT)

FIGURE 4. COWL FLAPS INSTALLATION

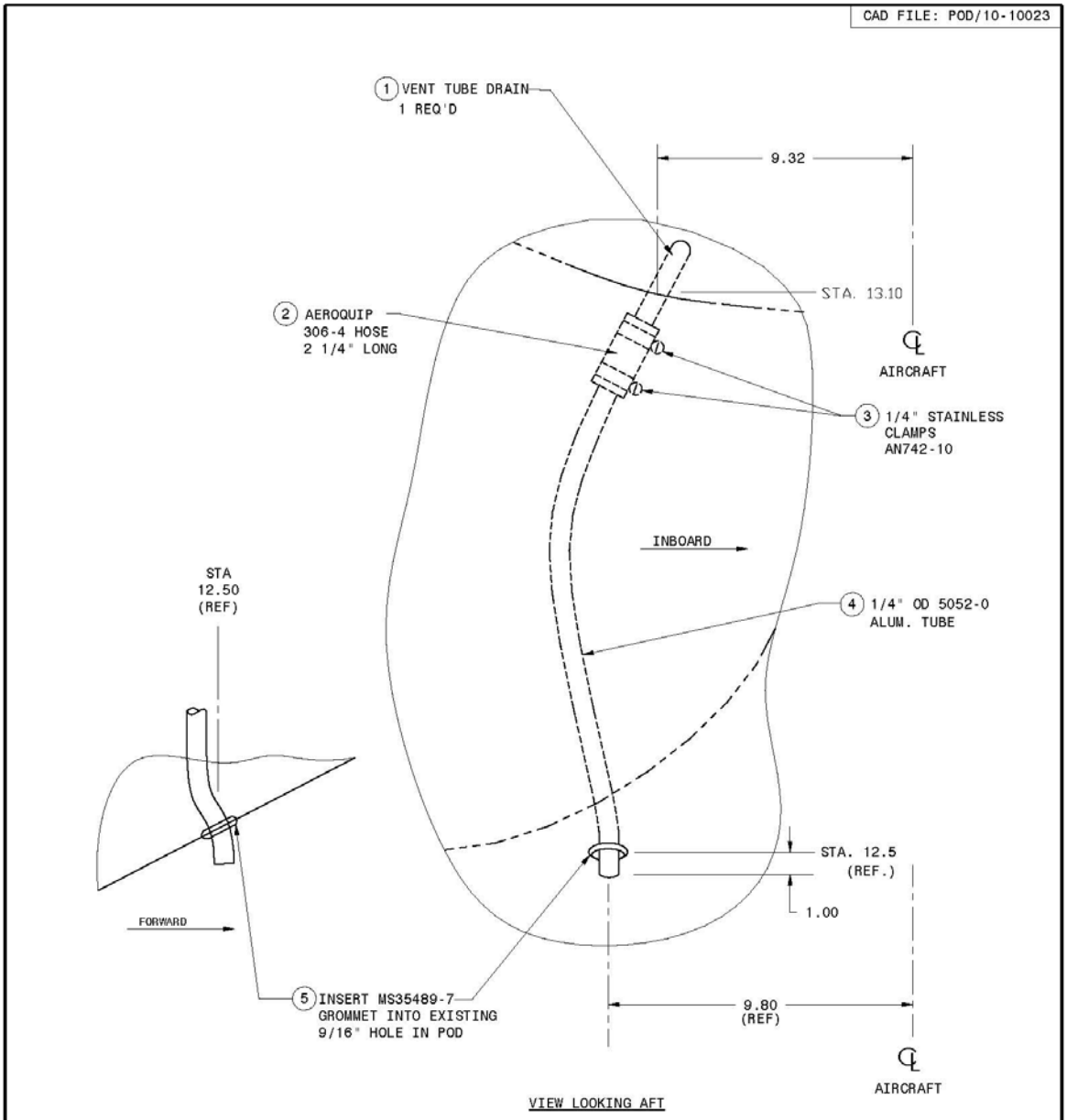
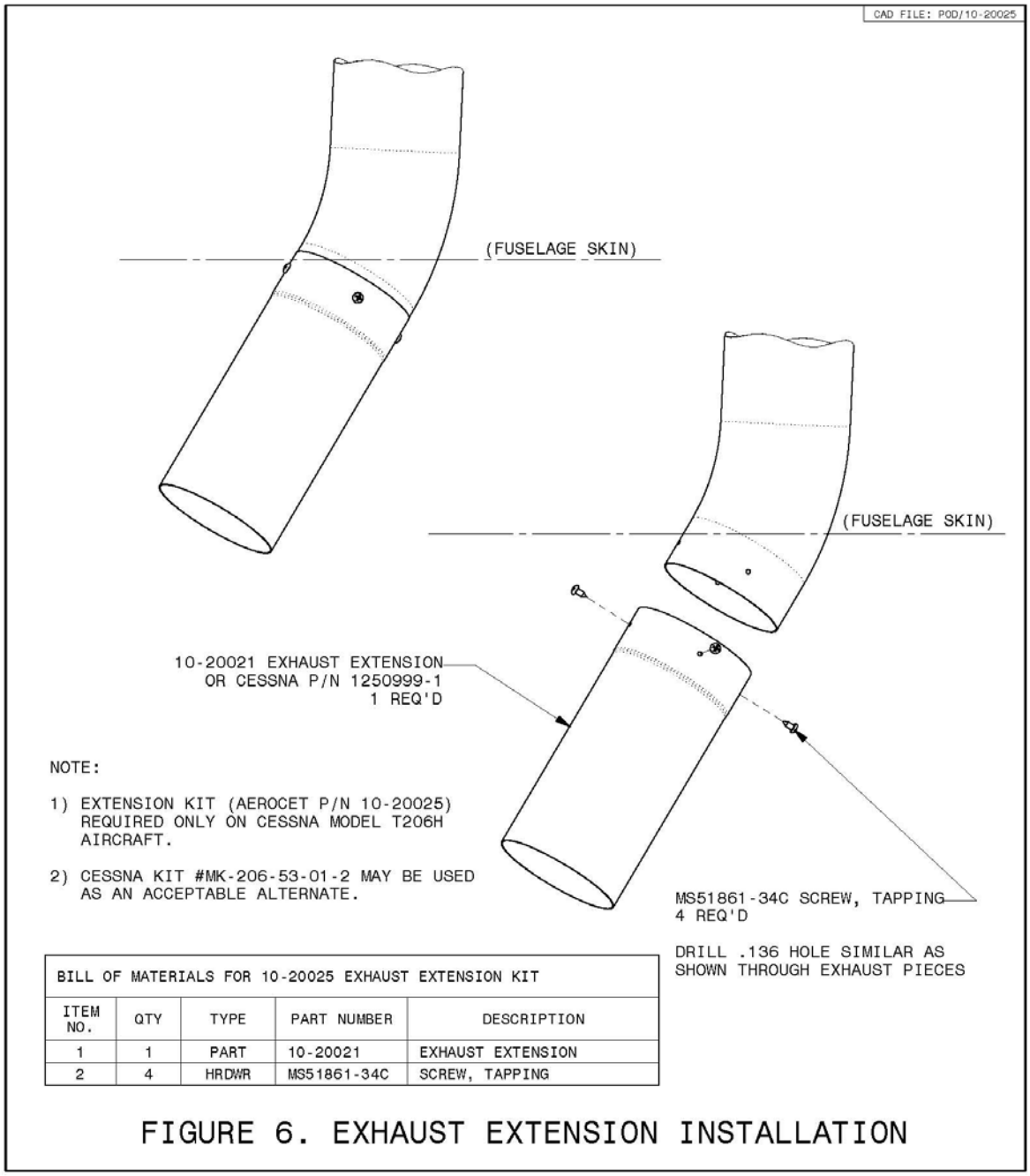


FIGURE 5. AUXILIARY PUMP DRAIN MODIFICATION


CAD FILE: POD/10-20025
(FUSELAGE SKIN)
(FUSELAGE SKIN)
10-20021 EXHAUST EXTENSION
OR CESSNA P/N 1250999-1
1 REQ'D
NOTE:
1) EXTENSION KIT (AEROCET P/N 10-20025) REQUIRED ONLY ON CESSNA MODEL T206H AIRCRAFT.
2) CESSNA KIT #MK-206-53-01-2 MAY BE USED AS AN ACCEPTABLE ALTERNATE.
MS51861-34C SCREW, TAPPING
4 REQ'D
DRILL .136 HOLE SIMILAR AS SHOWN THROUGH EXHAUST PIECES

BILL OF MATERIALS FOR 10-20025 EXHAUST EXTENSION KIT

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-20021	EXHAUST EXTENSION
2	4	HRDWR	MS51861-34C	SCREW, TAPPING

FIGURE 6. EXHAUST EXTENSION INSTALLATION

FAA APPROVED

SUPPLEMENTAL AIRPLANE FLIGHT MANUAL

FOR

CESSNA MODEL 206H AIRPLANES

EQUIPPED WITH AEROCET 206 CARGO PACK

Registration No. _____

Serial No. _____

The information contained in this document is FAA approved material which must be applied together with the basic FAA approved airplane placards and markings and/or FAA approved Airplane Flight Manual. This supplemental manual must be carried in the airplane when it is modified by the installation of the Aerocet 206 Cargo Pack in accordance with Supplemental Type Certificate (STC) No. SA00096SE. The information contained in this document supersedes the basic airplane markings and placards and/or Flight Manual covered in the items contained herein. For Limitations, Procedures, and Performance information not contained in this supplement, consult the basic airplane markings and placards, and/or Flight Manual.

FAA Approved: _____
FOR Manager, Seattle Aircraft Certification Office

Date: 19 JUNE 2006

LOG OF REVISIONS PAGE

REV.	PAGES AFFECTED	DESCRIPTION	FAA APPROVED	DATE

SECTION 1. GENERAL (206 CARGO PACK)

The cargo pack provides additional cargo and baggage space. The basic shell of the cargo pack, including the loading door, is fabricated from a fiberglass composite sandwich construction. A loading door is located on the left side of the pack, and is hinged at the bottom. It is secured in the closed position by three quick-release fasteners.

The volume of the cargo pack is 16 cubic feet. Dimensions of the pack and its loading door opening are contained in Section 6 of the basic Pilots Operating Handbook.

The pack is attached to the bottom of the fuselage with screws and, after the initial installation, can be readily removed or installed. Complete instructions for installation of the cargo pack, and required modifications to the nose gear access panels, fuel pump vent line, and cowl flaps, are contained in the Aerocet 206 Cargo Pod Installation Manual (Doc. A-10014).

This supplemental manual, applicable to those Cessna Model 206H airplanes equipped with the Aerocet 206 Cargo Pack, provides information and limitations not included in the basic FAA approved markings and placards, and/or Airplane Flight Manual. The aircraft is to be operated under the "NORMAL CATEGORY" only.

SECTION 2. LIMITATIONS

The ADF bearing accuracy may be adversely affected by the type and/or arrangement of the cargo pack contents.

The following information must be presented in the form of a placard, located on the inside of the cargo pack door. The fuel sump draining information placard is to be located on the outside of the cargo pack door.

REFER TO WEIGHT & BALANCE DATA
FOR BAGGAGE / CARGO LOADING
NEVER EXCEED 300 LBS CARGO WEIGHT.

10-20030

THE ADF BEARING ACCURACY MAY BE ADVERSELY AFFECTED BY THE TYPE AND / OR ARRANGEMENT OF THE CARGO PACK CONTENTS.

10-20017-2

FUEL SUMP DRAINS:
TWO AT STA. 23.40
DRAIN BEFORE FIRST FLIGHT
OF EACH DAY, OR AFTER REFUELING

10-20022

FAA Approved: _____

NOTE: The above fuel sump placard which is required to be located on the cargo pack door refers to the two drains located in the cargo pack. In addition as per normal procedures, before the first flight of the day and prior to flight following exposure to rain, sleet, snow, or refueling from an unfiltered fuel source: Drain and catch a sample from the fuel strainer, wing and belly sumps, and check for water.

SECTION 3. EMERGENCY PROCEDURES

There is no change to the airplane emergency procedures when the cargo pack is installed.

SECTION 4. NORMAL PROCEDURES

(NOTE: THESE ITEMS SUPPLEMENT THE CESSNA NORMAL PROCEDURES – BE SURE TO FOLLOW THE CESSNA PROCEDURES EXCEPT AS NOTED BELOW.)

a. NORMALLY ASPIRATED AIRCRAFT

Because of the effect of the cargo pack on climb performance, the maximum flap deflection approved for takeoff is 10° whenever the airplane is operated at weights above 3450 pounds. When operated at or below this weight, up to 20° of flap may be used.

At takeoff weights above 3450 pounds, short field takeoffs with 10° flaps should be conducted using a speed of 68 KIAS at the 50-foot obstacle. At weights of 3450 pounds or less, the speeds on the Takeoff Distance chart in Section 5 of the Pilots Operating Handbook and a 20° flap setting may be used. All other speeds remain unchanged from those listed in Section 4.

SECTION 5. PERFORMANCE

a. NORMALLY ASPIRATED AIRCRAFT

To obtain takeoff performance of the airplane with a cargo pack installed and using 10° flaps, increase both ground roll and total distance over the 50-foot obstacle by 10% over that found in Section 5 of the Pilots Operating Handbook. This procedure is required only when operating at weights above 3450 pounds. When operating at or below this weight, use 20° flaps and takeoff data in section 5 of the Pilots Operating Handbook.

The climb performance of the airplane equipped with a cargo pack is approximately 40 ft/min less than that shown in the Rate of Climb chart for the standard airplane.

To obtain speed performance for the airplane equipped with a cargo pack, the cruise speeds are approximately 5 KTAS lower than shown in the cruise charts in Section 5 of the Pilots Operating Handbook.

FAA Approved: _____

SERVICE MANUAL

AND

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

FOR

AEROCET 206 CARGO PACK

This ICA must be followed when the Cargo Pack is installed in accordance with Supplemental Type Certificate, (STC) No. SA0096SE, dated June 7, 1994.

The information contained in this document supplements or supercedes the basic manual only in those areas listed herein. For limitations, procedures, and performance information not contained in this manual, consult the basic aircraft ICA or Maintenance Manual.

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Rev 0 was approved by FAA without signature sheet.

STATEMENT OF Rev.1 CERTIFICATION

This manual complies with Federal Aviation Association (FAA) Airworthiness Requirements Part 23.

FAA Acceptance: [signature] Date: 2/2/2012

The above certification does not apply to revisions or amendments made after the date of initial certification by other Approved Organizations. Revisions or amendments made by other Approved Organizations must be separately certified and recorded on separate record sheets

STATEMENT OF Rev 2 CERTIFICATION

The changes in revision C are classified as minor and do not affect the FAA approved sections of this manual. The changes to the IPC include data that has FAA design data approval.

LOG OF REVISIONS PAGE

REV.	PAGES AFFECT.	DESCRIPTION	DATE	APPROVED
0	ALL	Initial Release	4/20/06	
1	5	Changed "Continued Airworthiness…) to "Recommended Airworthiness…), Table of Contents, Section 4	7/01/10	
	5	Added Section 5, "Airworthiness Limitations", and Section 6, "Illustrated Parts Catalogue", to Table of Contents		
	7,8	Added sub-sections 1.1, "Availability", 1.2, "Component Maintenance Manuals", 1.3, "Installation Manual", 1.4, "Dimensions, Locations, and Nomenclature", Figures 1.4.1 and 1.4.2 to document		
	9-12	Added Sections 2.1, Fastener Torque, 2.2, Fastener Torque Values, and 2.3, Fastener Use and Discretion to document		
	14	Added "Warnings" 3x, beginning section 3.4, Resin Mixing		
	15	Added 2nd paragraph, "Gel times should be assessed…", Section 3.4, Resin Mixing		
	15	Added "(Obsolete – use Hydrex 100 33350, not shown)", at "Typical Gel Times (Hydrex 33-253)", table heading		
	15	Deleted "Warning", which was duplicate of 1 of 3 "Warnings" added previously		
	17	Changed Section 4 heading, "Recommended Airworthiness", was "Continued Airworthiness"		
	149	Added Section 5, "Airworthiness Limitations" to document		
	21-28	Added Section 6, "Illustrated Parts Catalogue", to document		
2	ALL 2 24-28	Rev 2 certification statement. New Figure showing washers on rivets for Fig. 6.1.3, 6.1.4, 6.1.5, 6.1.6.	3/22/12	

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SECTION 1. Introduction and General Cargo Pack Information

This service and maintenance manual is applicable to Aerocet 206 Cargo Pack and its general application to a variety of Cessna Model 206 airplanes.

The Aerocet 206 Cargo Pack is all composite construction and is mounted to the belly of the aircraft according to the approved installation document. (Reference the STC document for specific listing. At the time of this writing, Document A-10014 is applicable to all aircraft models.)

1.1 Availability

One complete hard (paper) copy of this manual shall be provided with each new 206 Cargo pack. Additional copies and minor revisions shall be available via email, U.S.P.S. (mail), UPS or FedEx by request. Fees and delivery charges may apply.

Notification of any changes that require service for airworthiness shall be distributed to all applicable Aerocet owners on record with Aerocet, Inc. In such a case, copies of the applicable, revised portions of this manual shall be provided.

Aerocet, Inc. maintains record of purchasers and/or owners, collected at the time of purchase in order to comply with the above, as well as to maintain a high standard of service. If you have moved since your original purchase, have purchased a used product or otherwise have reason to believe that the contact information on file is incorrect, please provide the following information to Aerocet, Inc: (Aerocet contact information is on the front of this document.)

Cargo Pod Information:

Cargo Pod S/N _____

Aircraft Information:

Aircraft Make/Model _____
Aircraft Registration _____
Aircraft S/N _____

Owner Information: (as applicable)

Previous Owner _____
Previous Address _____
Present Owner _____
Present Address _____
Present Phone Number _____
Present Email Address _____

1.2 Component Maintenance Manuals

Certain accessories that are more complex, that require additional inspection and maintenance procedures may be described in Component Maintenance Manuals (CMM) or Maintenance Manuals (MM). These documents would be provided as part of the documentation provided with your product. At the time of this writing, no such components are used with Aerocet 206 Cargo Pod.

This subsection is reserved in the case of such future use.

1.3 Installation Manual

Installation instructions are provided as a part of the documentation package provided with the original purchase or upon request as detailed above. Refer to Aerocet Document Number A-10014 for instructions and installation parts lists.

1.4 Dimensions, Locations and Nomenclature

(See A-10014, values are reference here)

BASIC SPECIFICATIONS:

Maximum Weight: 300 lbs.
Maximum Load: 200 lbs/ft^2
Internal Volume: 16 ft2
Overall Length: 97.5” (8’1 ½”)
Opening for Baggage: 10.1” x 27.1”
Width (along 30” of mid-section): 40.5”

CHANGE IN WEIGHT AND BALANCE:

Weight Increase: 35.0 pounds (net change)
Arm: 51.0 inches
Resultant Moment: 1785.0 pound-inches
Index: 1.632

Figure 1.4.1 Showing side view of Aerocet Cargo Pod on Cessna 206 Aircraft.

Figure 1.4.2 Showing front view of Aerocet Cargo Pod on Cessna 206 Aircraft.

SECTION 2. Cargo Pack Maintenance

Corrosion is minimized due to the composite construction. The cargo pack should be kept clean with soap and water. External stains may be removed using commercially available marine fiberglass stain remover. We have used a product called FSR with good success. These surfaces may also be waxed to help in the cleaning process.

-Do not use abrasive cleaners or pads – these will scratch the white gel-coat surface. The gel-coat color surface should always be maintained on external surfaces for ultraviolet protection.

Cargo door provides access for inspection to view internal damage and to comply with the maintenance inspections. (See Table 1 in Section 4 for scheduled maintenance. See also: "Exceptional Inspections".)

Any penetration to the shell structure, delamination of the cloth layers, or gelcoat having worn through must be repaired according to hull repair Section 3. Significant damage (i.e. damage caused by ground or animal strikes) may warrant consultation with Aerocet, Inc. Significant damage may be categorized as damage that penetrates beyond the outside laminates to the core and/or the inside laminates, that is greater than 4" in diameter, or that compromises attachment fasteners.

Latches and seals are to be maintained as needed for proper function.

If applicable, Exhaust Extensions, which are required only on Cessna T206H airplane applications, should be inspected for stress cracking, corrosion, damage and condition.

2.1 Fastener Torque

2.1.1 Torque Value Conversion:

To convert in.-lbs. to ft.-lbs: Value (ft.-lbs.) x 12 = Value (in-lbs.)
To convert ft.-lbs. to in.-lbs: Value (in-lbs.) x 0.0833 = Value (ft.-lbs.)

2.1.2 Tooling Requirements:

Calibrated torque wrench
Adapters that affect the length of the torque wrench will affect the required torque indication and must be calculated according to Figure 2.2.3.

2.1.3 Hardware Cleanliness:

All hardware is to be free of dirt, grit and grease. All dirty hardware shall be thoroughly cleaned and lubricated with a dry film lubricant such as LPS 1, or Teflon products per manufacturer instructions. It is recommended that all stainless hardware be thoroughly lubricated with anti-seize lubricant of good quality to prevent galling upon assembly.

2.1.4 Torque Procedure

Assure that hardware is clean and properly prepared for installation. Assemble nuts to bolts, measuring the tension required to turn the nut and add this to the required final torque. Where possible apply torque to the nut, and not to the fastener head. Apply a smooth, even pressure, stopping and re-torquing if chattering or premature loading occurs. This may warrant disassembly and subsequent inspection for burrs or galling. Replace any damaged hardware.

Cross – recessed, Machine – pan head style screws should be torqued only to "hand tight", the fiberglass should exhibit only mild deformation. A portable hand drill could be used, provided that the clutch is set properly. All other fasteners shall be torqued as shown in this section unless otherwise noted.

2.2 Fastener Torque Values
(Except where otherwise noted)

CAUTION
THE FOLLOWING TORQUE VALUES ARE DERIVED FROM OIL FREE CADMIUM PLATED THREADS.

	TORQUE LIMITS RECOMMENDED FOR INSTALLATION (BOLTS LOADED PRIMARILY IN SHEAR)		MAXIMUM ALLOWABLE TIGHTENING TORQUE LIMITS	
Thread Size	Tension type nuts MS20365 and AN310 (40,000 psi in bolts)	Shear type nuts MS20364 and AN320 (24,000 psi in bolts)	Nuts MS20365 and AN310 (90,000 psi in bolts)	Nuts MS20364 and AN320 (54,000 psi in bolts)
FINE THREAD SERIES				
8-36	12-15	7-9	20	12
10-32	20-25	12-15	40	25
1/4-28	50-70	30-40	100	60
5/16-24	100-140	60-85	225	140
3/8-24	160-190	95-110	390	240
7/16-20	450-500	270-300	840	500
1/2-20	480-690	290-410	1100	660
9/16-18	800-1000	480-600	1600	960
5/8-18	1100-1300	600-780	2400	1400
3/4-16	2300-2500	1300-1500	5000	3000
7/8-14	2500-3000	1500-1800	7000	4200
1-14	3700-5500	2200-3300*	10,000	6000
1-1/8-12	5000-7000	3000-4200*	15,000	9000
1-1/4-12	9000-11,000	5400-6600*	25,000	15,000

Figure 2.2.1 Recommended Torque Values (inch-pounds)
(From AC43.13-1B, Table 7-1)

FINE THREAD SERIES	
THREAD SIZE	**MINIMUM PREVAILING TORQUE**
7/16 - 20	8 inch-pounds
1/2 - 20	10 inch-pounds
9/16 - 18	13 inch-pounds
5/8 -18	18 inch-pounds
3/4 - 16	27 inch-pounds
7/8 - 14	40 inch-pounds
1 - 14	55 inch-pounds
1-1/8 - 12	73 inch-pounds
1-1/4 - 12	94 inch-pounds
COARSE THREAD SERIES	
THREAD SIZE	**MINIMUM PREVAILING TORQUE**
7/16 - 14	8 inch-pounds
1/2 - 13	10 inch-pounds
9/16 - 12	14 inch-pounds
5/8 - 11	20 inch-pounds
3/4 - 10	27 inch-pounds
7/8 - 9	40 inch-pounds
1 - 8	51 inch-pounds
1-1/8 - 8	68 inch-pounds
1-1/4 - 8	88 inch-pounds

Figure 2.2.2 Minimum Prevailing Torque Values for Re-used Self-Locking Nuts
(from AC43.13-1B, Table 7-2)

Self-Locking Nuts:

Self-locking nuts, when re-used, must have at least the minimum prevailing torque listed in figure to the left. Nuts that are smaller than those listed in the table shall not be used if they can be run up by hand.

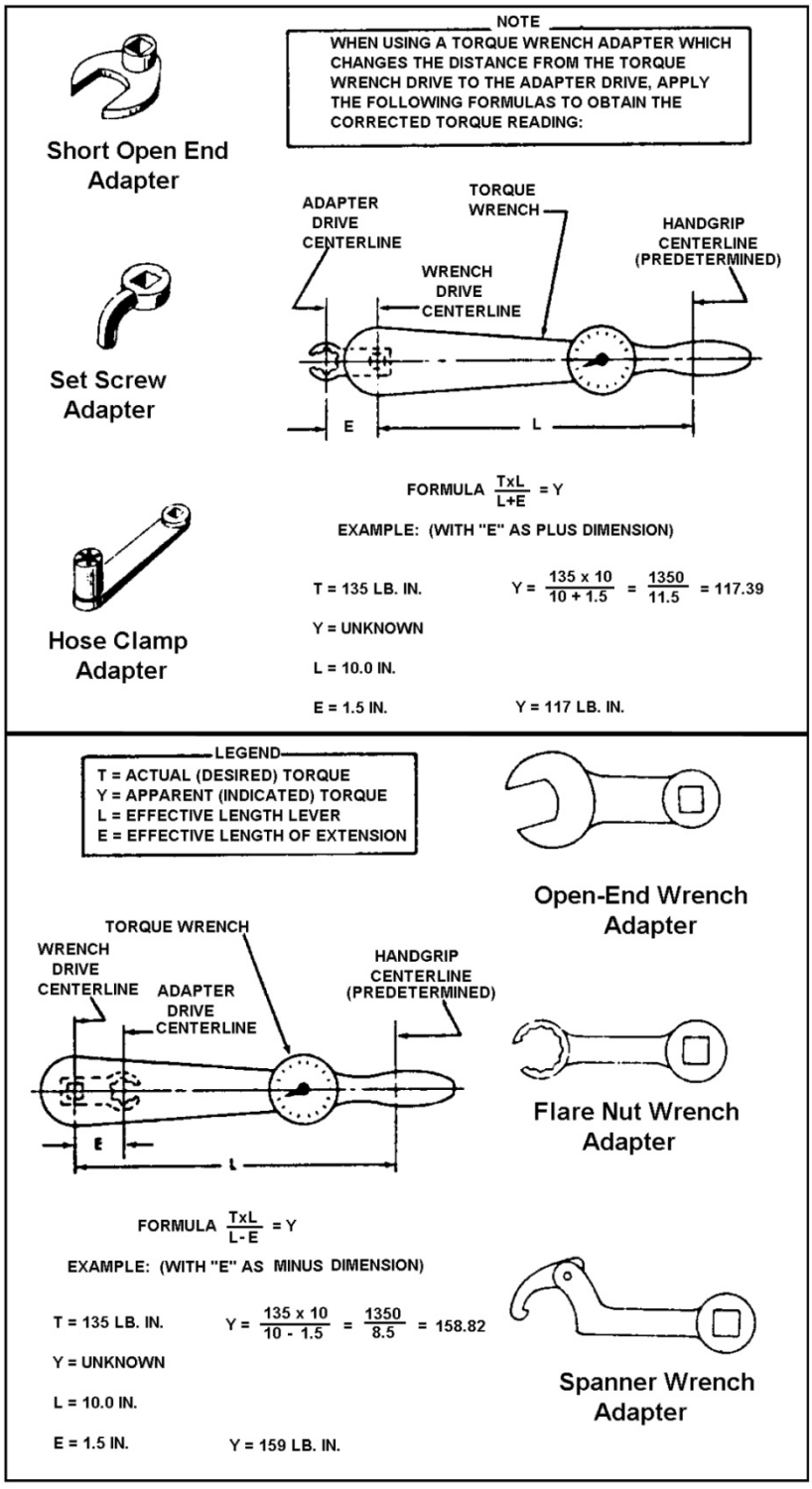
NOTE
WHEN USING A TORQUE WRENCH ADAPTER WHICH CHANGES THE DISTANCE FROM THE TORQUE WRENCH DRIVE TO THE ADAPTER DRIVE, APPLY THE FOLLOWING FORMULAS TO OBTAIN THE CORRECTED TORQUE READING:
Short Open End Adapter
Set Screw Adapter
Hose Clamp Adapter
ADAPTER DRIVE CENTERLINE
TORQUE WRENCH
WRENCH DRIVE CENTERLINE
HANDGRIP CENTERLINE (PREDETERMINED)
E
L
FORMULA TxL / L+E = Y
EXAMPLE: (WITH "E" AS PLUS DIMENSION)
T = 135 LB. IN.
Y = UNKNOWN
L = 10.0 IN.
E = 1.5 IN.
Y = 135 x 10 / 10 + 1.5 = 1350 / 11.5 = 117.39
Y = 117 LB. IN.
LEGEND
T = ACTUAL (DESIRED) TORQUE
Y = APPARENT (INDICATED) TORQUE
L = EFFECTIVE LENGTH LEVER
E = EFFECTIVE LENGTH OF EXTENSION
Open-End Wrench Adapter
TORQUE WRENCH
WRENCH DRIVE CENTERLINE
ADAPTER DRIVE CENTERLINE
HANDGRIP CENTERLINE (PREDETERMINED)
E
L
Flare Nut Wrench Adapter
FORMULA TxL / L-E = Y
EXAMPLE: (WITH "E" AS MINUS DIMENSION)
T = 135 LB. IN.
Y = UNKNOWN
L = 10.0 IN.
E = 1.5 IN.
Y = 135 x 10 / 10 - 1.5 = 1350 / 8.5 = 158.82
Y = 159 LB. IN.
Spanner Wrench Adapter

Figure 2.2.3 Torque Wrench with Various Adapters
(from AC43.13-1B, Figure 7-2)

2.3 Fastener Use and Discretion

2.3.1 Fastener Lengths

Rivets: Where replacement or repair of rivets is required, use rivets of proper specifications only. For instance, MS203426AD4-xxx. Lengths may be determined by measuring the thickness of the material(s) to be assembled and adding 1.5 X Diameter of the rivet to be used. Over-sized rivets may be substituted where holes have been drilled out.

Bolts and screws shall have a minimum of one thread visible through the nuts upon final torque.

Washers may be rearranged if necessary to accommodate proper fit, up to two washers beneath the nut and one beneath the fastener head. Typically, Aerocet intends to put one thin washer beneath the fastener head and one thicker washer beneath the nut.

2.3.2 Fastener Reuse

Fasteners are to be inspected for condition, per Section 6 of this manual. Such fasteners that are acceptable may be cleaned, re-lubricated and re-installed as determined. Self-locking nuts shall meet the minimum prevailing torque as listed in Figure 2.2.2, or shall be replaced.

SECTION 3. Repairing Composite Structure

3.1. General Description

Composite structure repair, done correctly, will obtain the strength required to return the cargo pack to normal service and show little or no cosmetic evidence of damage having taken place. The materials used for original construction and repair are conventional to the industry. Contact Aerocet, Inc. prior to beginning a repair to obtain correct materials - resin (resins have shelf lives), catalyst, cloth, gel-coat, resin thickeners. "In-the-field" epoxy patch kits may be used in an emergency if the damage is relatively small, but the repair must be replaced with correct materials for equal strength status. Damage larger than 4" in size requires consulting Aerocet, Inc. for proper laminate orientation and counts in the damaged area.

3.2. Repair Types and Procedures – All repaired areas on the exterior must be surface coated (gel-coated) with a minimum of 10 mil thickness to assure UV protection.

3.2.1. **Resin Starved Areas, Exposed Fibers, or Small Impact Damaged Soft Spots (1/2" Dia. or less).**

3.2.1.1. If necessary sand surfaces to remove gloss.

3.2.1.2. Use brush, squeegee, or hypodermic to work resin into defective area. (Use same resin as the original laminate).

3.2.2. **Small Bruises, Punctures Less than 1/4" diameter, or Surface Voids.**

3.2.2.1. Sand surface surrounding defect to remove gloss.

3.2.2.2. Use same fabric as the original part.

3.2.2.3. Cut patches to fit correction area, extending a minimum of 1/2" past the damaged area. All patch corners must be rounded.

3.2.2.4. Apply a light brush coat of resin (similar to original).

3.2.2.5. Place one or more plies on detail covering correction area using impregnation of fabric as described in section 3.3.

3.2.3. **Cuts, Fractures, or Punctures 1/4" diameter or Larger.**

3.2.3.1. Cut back material as required to ascertain extent of damage. Trim plies outward to a smooth oval (1/2"per ply generally).

3.2.3.2. If the area is large enough supply backing to hold the shape of the original contour. Put a parting agent on this backing to assure its release.

3.2.3.3. Replace the fabric on a ply-for-ply basis overlapping 1/2" minimum on each succeeding ply using impregnation of fabric as described in section 3.3. (Any smooth areas need to be sanded with 80 grit sandpaper to assure proper bonding.)

3.2.3.4. If damage has occurred where there is sandwich construction and involving the core, then work each layer separately. Fix either the outside or inside skins. Then cut to fit like core material to replace the damaged core. Bond the core onto the repaired skin using the proper resin and thickener. A mixture of Hydrex resin and Aerosil 202 thickener should be applied to the bonding surface of the core using a squeegee (Torin Corebond alt. OK). A film of approximately .015" should be used. Pressure needs to be applied for the bond to assure proper adhesion to the skin eliminating air voids. This pressure can be applied to small area core bonding by weights (ex. lead shot bags with a release film to keep from sticking to any excess). Larger areas require core bonding using a vacuum bag. Consult Aerocet, Inc. for this procedure. Fill any seam voids with a resin/glass bubble mixture. Apply the final laminates accordingly to finish the repair.

3.3. Impregnation of Fabric

Resin impregnation can be accomplished by laying cloth, cut to a suitable pattern, on a flat table and applying resin mixture with a squeegee so as to achieve an even impregnation. Similarly, it may also be done directly on the part with the cloth being applied with either a squeegee or a brush, assuring that voids and starved areas are not produced. It is very helpful to apply a thin coat of resin to the area to be laminated. Then lay the cloth down, rolling the cloth into the resin. Any air in the laminate should be squeegeed or brushed out. See section 8.4 regarding resin mixing.

3.4. Resin Mixing

WARNING

Observe all safety regulations and recommendations of the manufacturer according to the Material Safety Data Sheet (MSDS) and/or other instructions for the safe use and handling of all resin related products.

WARNING:

Cobalt should never be mixed directly with MEKP catalyst. A violent reaction will occur which may result in fire or explosion. Cobalt should be stored separately from the catalyst.

WARNING:

Be careful with the MEKP catalyst. Contact with eyes must be prevented. Flush eyes immediately, contact a physician immediately. Never mix MEKP without eye protection. Read recommendations on catalyst supply bottle.

Gel time or pot life is the time it takes the resin to set up in the container after proper and thorough mixing with accelerators and catalysts. Gel times can be adjusted significantly by varying the amounts of these materials. Gel times also will vary significantly with the batch size if left in a bucket or with a very thick laminate.

Gel times should be assessed with each new batch or type of resin system used in the working environment. Ambient air and resin temperatures, shelf-life and catalyst ratios may affect the gel times significantly. As previously stated, thicker laminates and/or resin pots will cure more quickly due to accumulation of heat during gel process. Thin laminates such as specified for use in many Aerocet products will often gel in double the amount of time needed to gel a 100g in a 4 in. diameter pot.

Resin catalysis shall not be accomplished in ambient air temperatures of less than 50°F, except under extreme circumstances for the purposes of a temporary field repair. In no case shall more than 2% catalyst mixture be used; and in no case shall less than the recommended (.75%) catalyst mixture be used.

TYPICAL GEL TIMES (HYDREX 33-253) *(Obsolete – Use Hydrex 100 33350, not shown)*100g castings only, laminate times typically double

33%MEKP %Catalyst	Resin Qty	50°F	60°F	70°F	80°F
.75%	100gm	55 min	38 min	28 min	23 min
1.00%	100gm	40 min	27 min	20 min	15 min
1.50%	100gm	30 min	21 min	14 min	11 min
2.00%	100gm	23 min	17 min	12 min	8 min

Note: *Under no circumstances* should more than 2.0% catalyst mixture be used. Also, do not use less than the recommended minimum amount of catalyst (.75%) or the resin may never completely cure, resulting in a reduction of strength.

3.5. Preparation of Fiberglass Materials

3.5.1. Fiberglass shall be trimmed on a clean table to prevent contamination.

3.5.2. When laps are necessary, lap widths of at least 1/2" shall be maintained for fiberglass pieces in any given ply and no more than one of the component plies shall be lapped at any one place. The number of laps shall be kept to a minimum.

3.6. Surface Coat Application (Gel Coat)

3.6.1. All surface coats must be applied to a thickness of 10 to 15 mil. Use a mil gauge and check often.

3.6.2. All surface coats must be catalyzed with 2% MEKP.

3.6.3. Thinning of surface coats can only be done to manufacturers recommendations.

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SECTION 4. Recommended Airworthiness Service Schedule, General Practices and Product Listings for Service

4.1. Recommended Service Schedule – Table 1

TABLE 1

INSPECTION TIME INTERVALS		HOURS/(MONTHS) (Whichever occurs first)			
	Notes	25	50	100 (12)	200 (24)
Exterior	Inspect for damage, surface coat (gel coat - UV protection).	X			
Interior	Inspect for evidence of damage.			X	
Cargo Door	Inspect seals, latches and hinges for proper function, corrosion and damage.				X
Hardware	Spray coat protection according to Note #1, Hardware according to Note #2 (if desired).			X	
Placards	Inspect for placement & legibility. (See installation manual for proper locations).				X
Exhaust Extensions	Inspect for stress cracking and corrosion (applicable to Cessna T206H only).			X	

Note #1 – Spray coat of a migrating corrosion material (ACF-50, Boeshield T9, or Corrosion X).

Note #2 – Coat hardware with PUR-AL-KETONE or LPS 3.

4.2. "Exceptional" Inspections:

In a variety of circumstances, it is necessary to perform prompt inspections for damage. Details relating to these investigations are addressed in Sections 2 & 3 of this manual and Table 1 above. It is the responsibility of the pilot to determine the severity of damage, and the flightworthiness of the aircraft in the field. Inspections and repairs are to be performed as necessary and per practices outlined throughout this manual.

A list of possible scenarios includes, but is not limited to the following:

4.2.1 Harsh landings, especially when heavy cargo is loaded or has shifted in a significant manner.

4.2.2 Impact with an object during taxi, take-off or landing.

4.2.3 Suspected damage incurred during tie-down.

4.2.4 Visible damage found during normal preflight inspection.

4.3. General Practices

4.3.1 Metal Parts – check for corrosion, stress cracks or metal distortion, elongation of holes, rivet damage

4.3.2 Critical Bolts – check for corrosion (rust), wear, torque

4.3.3 Composite Parts – check for stress cracks, gel coat presence (UV protection), punctures

4.4. **Product Listings** (our recommendations)

4.4.1 Rust (corrosion) Protection – ACF-50, Corrosion X, or Boeing Co. Boeshield T9

4.4.2 Bolt Protection - PUR-AL-KETONE (Distributed by Lake & Air), LPS Industries LPS 3, Zip Chemical Co. Zip D-5029NS

4.4.3 Composite Materials for Hull Repair – Contact Aerocet, Inc. for resin, cloth, gel-coat, resin thickeners

Section 5. Airworthiness Limitations

5.1 GENERAL

The Airworthiness Limitations section is FAA approved and specifies inspections and other maintenance required under §§43.16 and 91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.

5.2 DESCRIPTION

1. **TIME LIMITED ITEMS.**

   None**.**

2. **REQUIRED INSPECTIONS INTERVAL.**

   None**.**

3. **SCHEDULED MAINTENANCE: None (**for Aerocet recommended Inspection or Replacement, refer to Section 4.

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Section 6. Illustrated Parts Catalogue

Section 6 contains part listings for the Aerocet 206 Cargo Pod. For installation – related part listings, refer to Aerocet installation manual A-10014.

6.1 LIST OF FIGURES

Figure 6.1.1. 206 Cessna Pod Door Versions

Figure 6.1.2. Parts Listing for 10-10010-1 Door Configuration

Figure 6.1.3. Parts Listing for 10-10010-1 Door Panel Assembly

Figure 6.1.4. Parts Listing for 10-10010-2 Door Configuration

Figure 6.1.5. Parts Listing for 10-10010-2 Door Panel Assembly

Figure 6.1.6. Parts Listing for 10-10010-3 Door Configuration

Figure 6.1.7. Parts Listing for 10-10010-3 Door Panel Assembly

Cargo Pod Door Versions

10-10010-1 Original Door design, utilizing Camloc latches and three bottom-positioned door hinges. Available for repair and replacement parts only, once -2 and -3 versions are available.

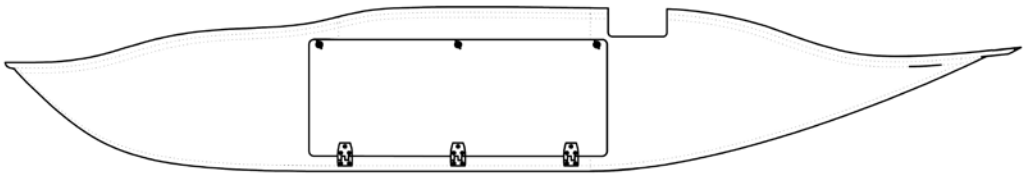
(Refer to Figures 6.1.2 and 6.1.3 for exploded views and part lists.)

10-10010-2 New Door design, utilizing Aerocet developed rotational latches and a piano hinge along the bottom edge. An available option on all new 206 Pod orders.

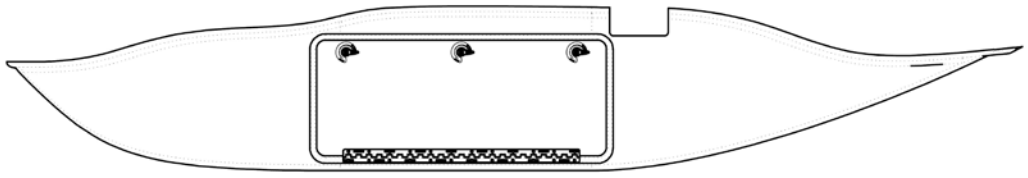
(Refer to Figures 6.1.4 and 6.1.5 for exploded views and part lists.)

10-10010-3 New Door design, utilizing Aerocet developed rotational latches and a piano hinge along forward edge. An available option on all new 206 Pod orders.

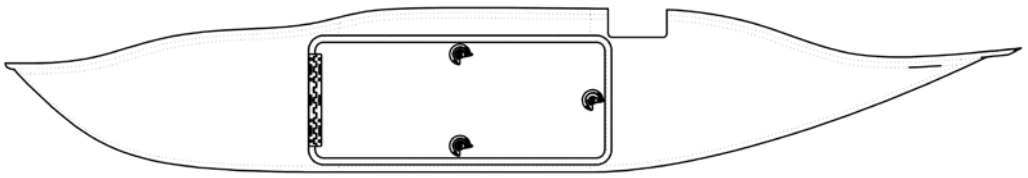
(Refer to Figures 6.1.6 and 6.1.7 for exploded views and part lists.)



10-10010-1 BOTTOM HINGING (ORIGINAL) VERSION

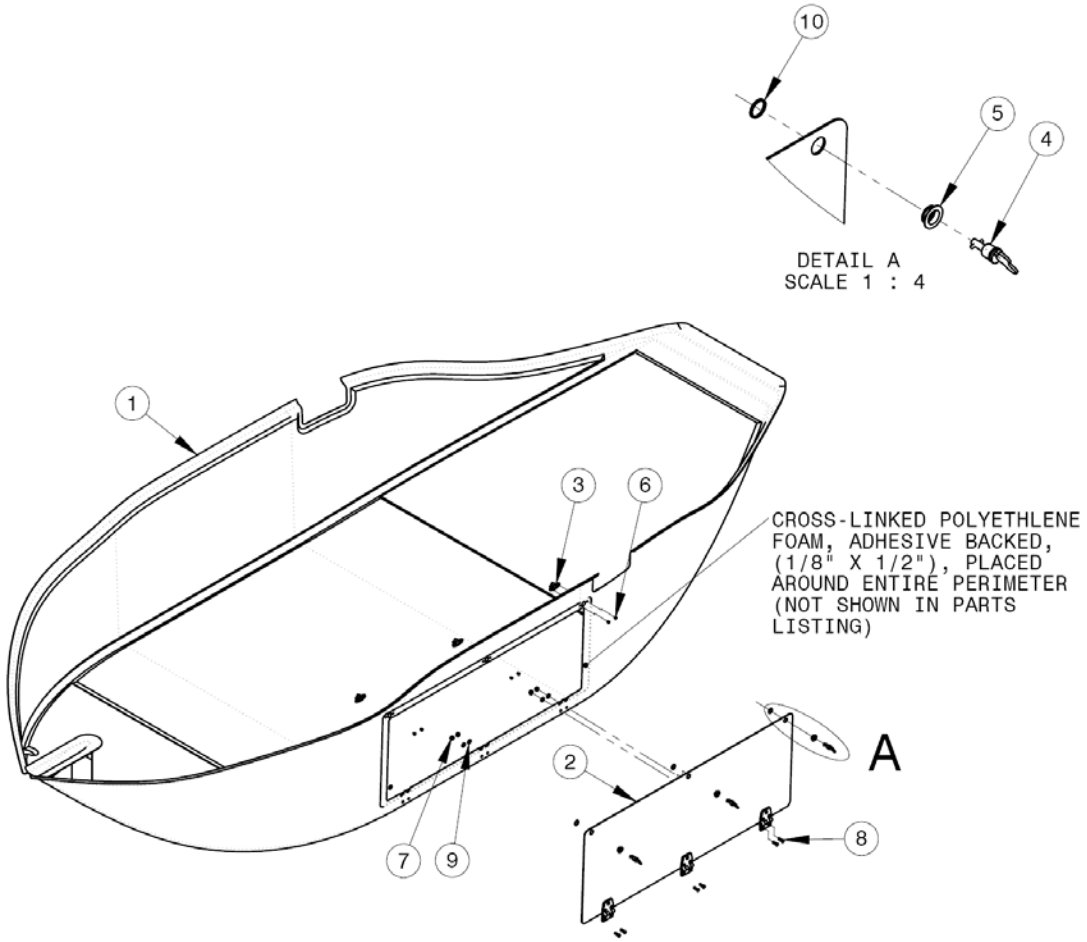


10-10010-2 BOTTOM HINGING (NEW) VERSION



10-10010-3 FORWARD HINGING (NEW) VERSION

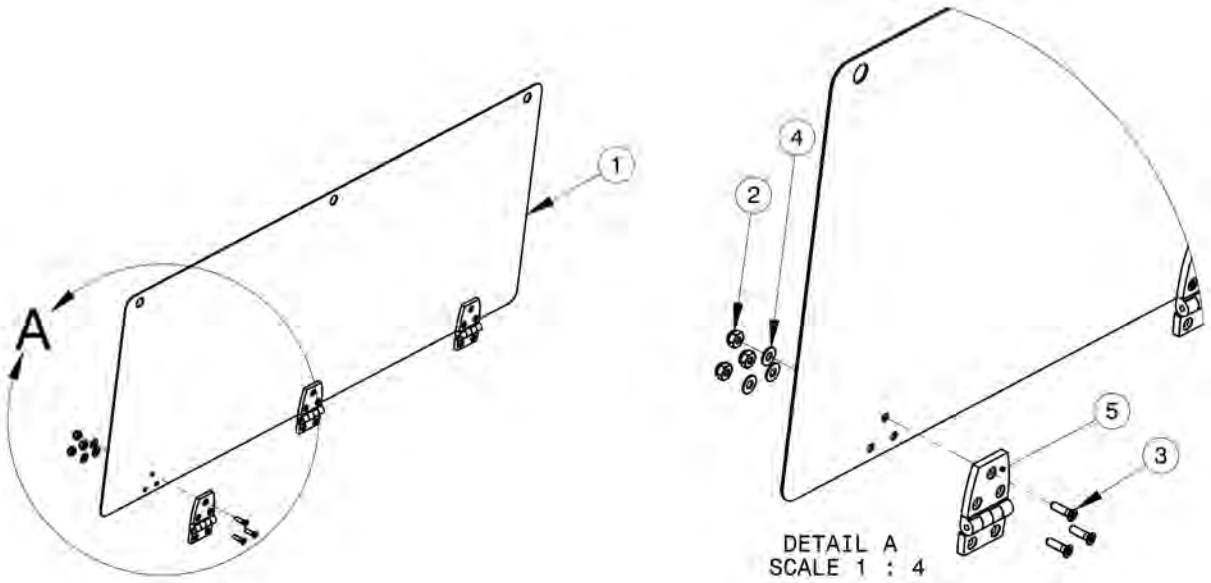
Figure 6.1.1. 206 Cessna Pod Assembly Versions



10-10010-1 POD ASSEMBLY, BOTTOM HINGING DOOR, ORIGINAL CONFIGURATION

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10010-11	AEROCET 206 CARGO POD SHELL, -1 OPTIONS
2	1	ASSY	10-10015	DOOR PANEL ASSEMBLY
3	3	PART	214-16N	CAMLOCK END
4	3	HRWDR	4002-7W	STUD, CAMLOC P/N
5	3	PART	4002-N	GROMMET
6	6	HRDWR	MS20426AD4-4	RIVET, 100° COUNTERSUNK HEAD, LENGTH A/R
7	6	HRDWR	MS21044C08	NUT, SELF-LOCKING, REGULAR HEIGHT, STAINLESS
8	6	HRDWR	MS24694C6	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
9	6	HRDWR	NAS1149C0332R	WASHER, FLAT, STAINLESS
10	3	HRDWR	R4G	SNAP RING, CAMLOC P/N

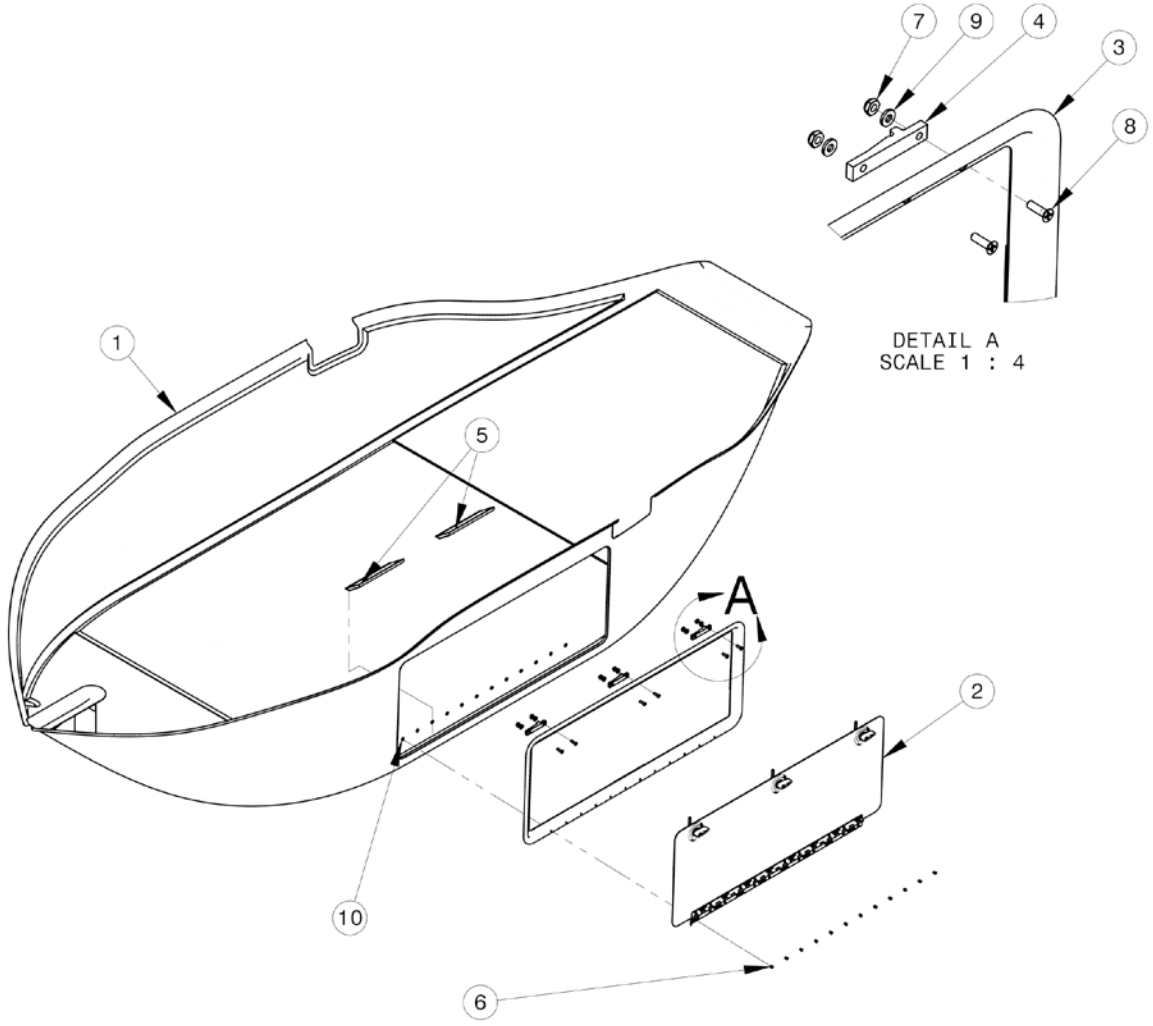
Figure 6.1.2. Parts Listing for 10-10010-1 Door Configuration



10-10015 ORIGINAL DOOR ASSEMBLY

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10016	DOOR PANEL
2	9	HRDWR	MS21044C08	NUT, SELF-LOCKING, REGULAR HEIGHT, STAINLESS
3	9	HRDWR	MS24694C6	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
4	9	HRDWR	NAS1149C0332R	WASHER, FLAT, STAINLESS
5	3	HRDWR	SD 202511-1	DOOR HINGE

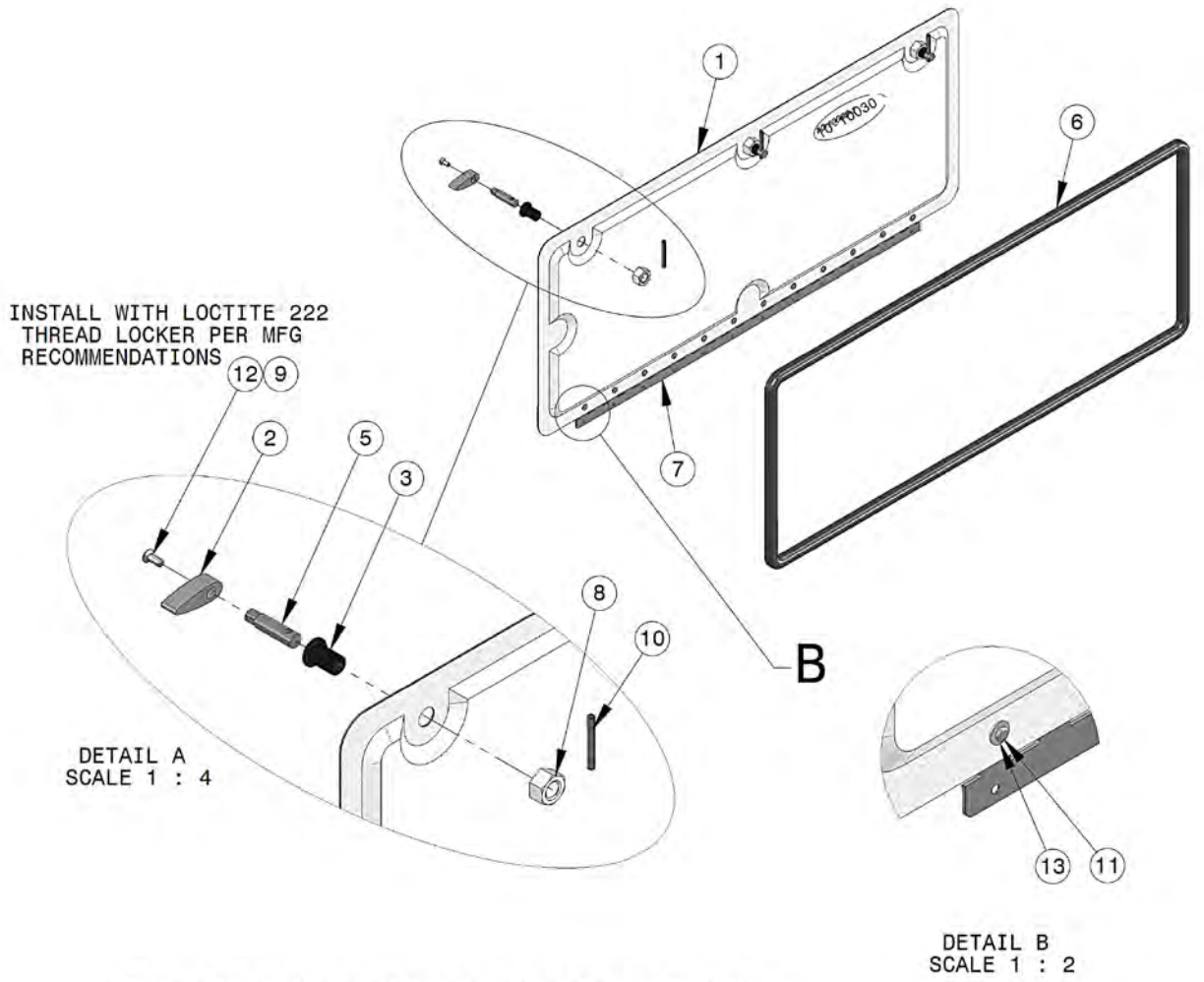
Figure 6.1.3. Parts Listing for 10-10015 Door Assembly



10-10010-2 POD ASSEMBLY, BOTTOM HINGING DOOR CONFIGURATION

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10010-21	AEROCET 206 CARGO POD SHELL, -2 & -3 OPTIONS
2	1	ASSY	10-10030-1	DOOR PANEL ASSEMBLY
3	1	PART	10-10038	TRIM RING AND DOOR INSET
4	3	PART	10-10051	LATCH RAMP
5	2	PART	10-10053	CARGO RAMP
6	12	HRDWR	MS20426AD4-5	RIVET, 100° COUNTERSUNK, LENGTH A/R
7	6	HRDWR	MS21083C3	NUT, SELF-LOCKING, LOW HEIGHT, STAINLESS
8	6	HRDWR	MS24694C52	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
9	6	HRDWR	NAS1149C0363R	WASHER, FLAT, STAINLESS
10	12	HRDWR	NAS1149FN432P	WASHER, FLAT

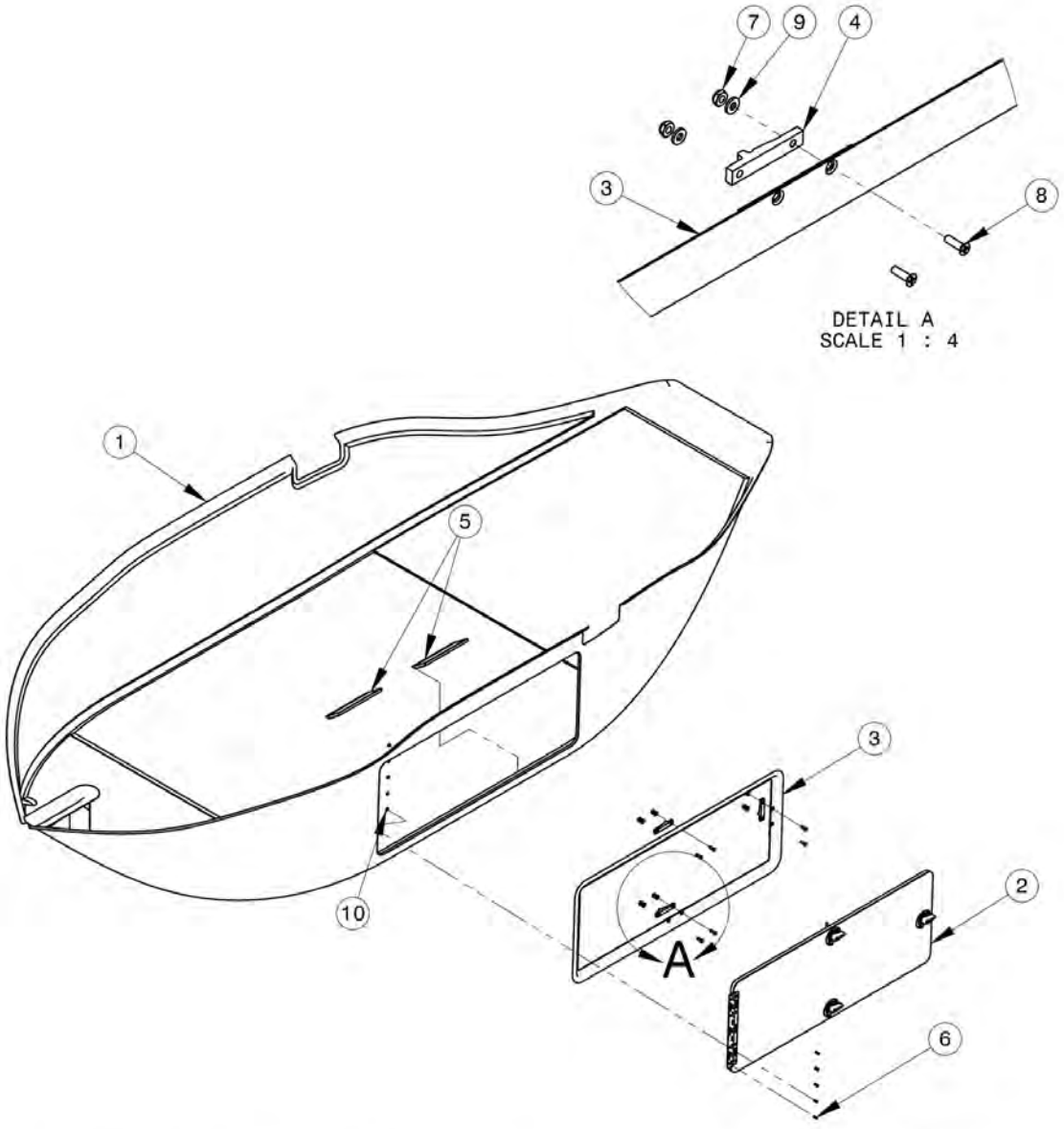
Figure 6.1.4. Parts Listing for 10-10010-2 Door Configuration



10-10030-1 BOTTOM HINGING DOOR ASSEMBLY

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10031	DOOR PANEL, BOTTOM HINGING
2	3	PART	10-10032	LATCH HANDLE
3	3	PART	10-10034	LATCH BUSHING
4	3	PART	10-10035	PANEL LATCH POSITION PLACARD
5	3	PART	10-10037	LATCH POST
6	1	HRDWR	10-10039	EPDM FOAM RUBBER SEAL, 1/2" D PROFILE
7	1	ASSY	10-10040-1	HINGE ASSEMBLY, AEROCET 206 CARGO POD
8	3	PART	35-80004	NUT, HEX, NYLON, 1/2-20 THREAD
9	1	HRDWR	LOCTITE 222	THREAD LOCKER
10	3	HRDWR	MS16562-240	SPRING PIN
11	12	HRDWR	MS20426AD4-5	RIVET, COUNTERSUNK (LENGTH A/R)
12	3	HRDWR	MS24694C49	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
13	12	HRDWR	NAS1149FN432P	WASHER, FLAT

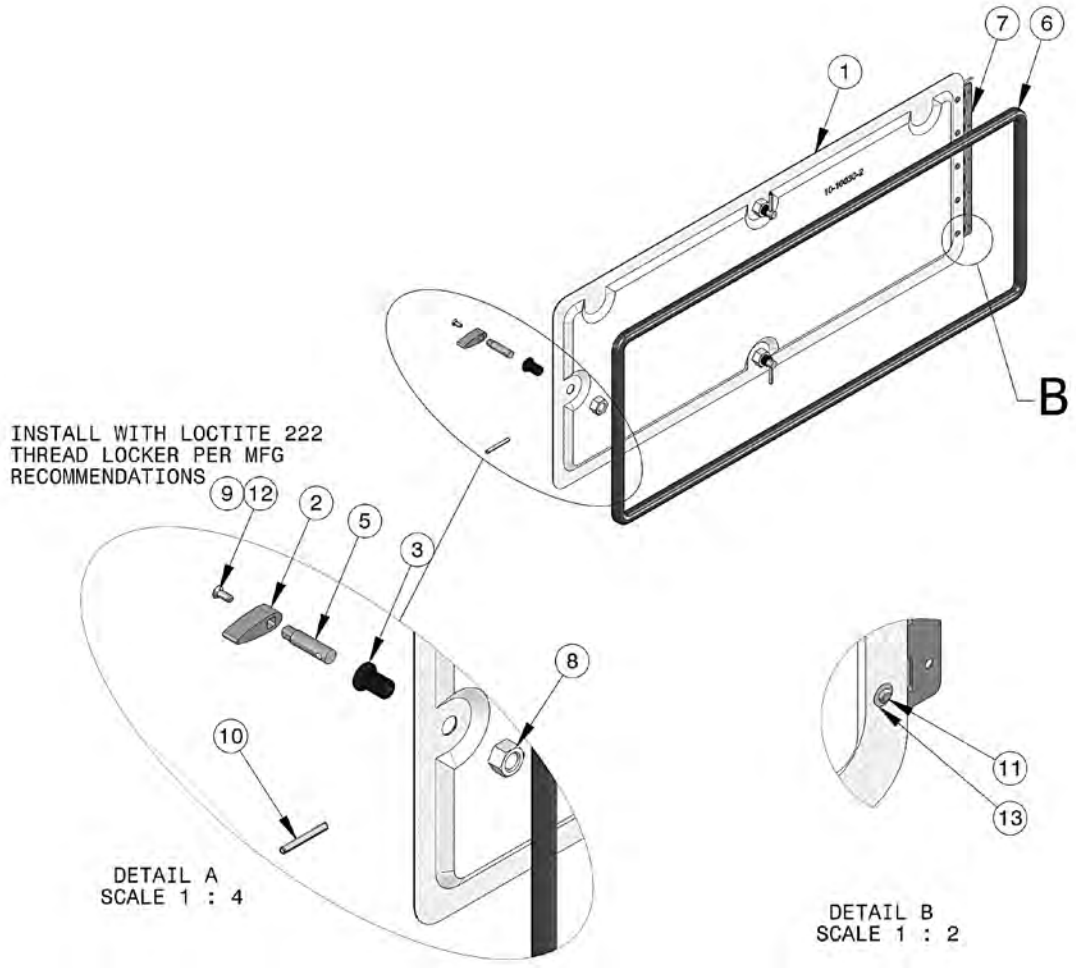
Figure 6.1.5. Parts Listing for 10-10030-1 Bottom Hinging Door Assembly



10-10010-3 POD ASSEMBLY, FORWARD HINGING DOOR CONFIGURATION

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10010-21	AEROCET 206 CARGO POD SHELL, -2 & -3 OPTIONS
2	1	ASSY	10-10030-2	DOOR PANEL ASSEMBLY, FORWARD HINGING
3	1	PART	10-10038	TRIM RING AND DOOR INSET
4	3	PART	10-10051	LATCH RAMP
5	2	PART	10-10053	CARGO RAMP
6	5	HRDWR	MS20426AD4-9	RIVET, COUNTERSUNK (LENGTH A/R)
7	6	HRDWR	MS21083C3	NUT, SELF-LOCKING, LOW HEIGHT, STAINLESS
8	6	HRDWR	MS24694C52	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
9	6	HRDWR	NAS1149C0363R	WASHER, FLAT, STAINLESS
10	5	HRDWR	NAS1149FN432P	WASHER, FLAT

Figure 6.1.6. Parts Listing for 10-10010-3 Door Configuration



10-10030-2 FORWARD HINGING DOOR ASSEMBLY

ITEM NO.	QTY	TYPE	PART NUMBER	DESCRIPTION
1	1	PART	10-10031	DOOR PANEL, FWD HINGING
2	3	PART	10-10032	LATCH HANDLE
3	3	PART	10-10034	LATCH BUSHING
4	3	PART	10-10035	PANEL LATCH POSITION PLACARD
5	3	PART	10-10037	LATCH POST
6	1	HRDWR	10-10039	EPDM FOAM RUBBER SEAL, 1/2" D PROFILE
7	1	ASSY	10-10040-2	HINGE ASSEMBLY, AEROCET 206 CARGO POD
8	3	PART	35-80004	NUT, HEX, NYLON, 1/2-20 THREAD
9	1	HRDWR	LOCTITE 222	THREAD LOCKER
10	3	HRDWR	MS16562-240	SPRING PIN
11	5	HRDWR	MS20426AD4-5	RIVET, COUNTERSUNK (LENGTH A/R)
12	3	HRDWR	MS24694C49	MACHINE SCREW, FLAT COUNTERSUNK HEAD, 100°
13	5	HRDWR	NAS1149FN432P	WASHER, FLAT

Figure 6.1.7. Parts Listing for 10-10030-2 Forward Hinging Door Assembly

End